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Volume I



MATERIAL CHARACTERIZATION, PART A MECHANICAL PROPERTIES OF  
TWO METALS AT SEVERAL STRAIN RATES

S. A. EMERY

University of Dayton Research Institute  
Dayton, Ohio 45469

May 1982

Interim Report for period 1 October 1977 - 30 November 1979

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of their use in the J-79 blade and the F-101 blade respectively. Part B of the report contains a discussion of mechanical property tests conducted on the two composite components of a hybrid composite blade: boron/2024 aluminum and stainless steel wire mesh. These two composites are used in the hybrid composite APSI blade.

Static, quasi-static and dynamic tensile tests were conducted on the metallic materials because (1) studies indicate that metallic mechanical properties exhibit strain rate dependence and (2) tests conducted on full scale blade, show that various locations on the fan blades load at different rates. The mechanical properties measured include Young's Modulus (when obtainable), Poisson's ratio (when obtainable), the yield strength, the ultimate strength, and the ultimate strain. The density of the materials was also measured.

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## FOREWORD

This report describes a contractual work effort conducted for the General Electric Company, Aircraft Engine Group under Purchase Order No. 200-FBA-14K-47844 which is a subcontract of F33615-77-C-5221.

This report covers work conducted during the period of October 1977 to November 1979 and is part of Task IV-A.

The GE Program Manager was Mr. Joe McKenzie and the Principal Investigator was Mr. Al Storace. The work reported herein was performed under the direction of Susan A. Emery, Experimental and Applied Mechanics Division, University of Dayton Research Institute.

Technical support was provided by Mr. E. C. Klein. Program management for the University was provided by Mr. Robert Bertke.

This report covers work conducted for project 3066, task 12, entitled Foreign Object Impact Design Criteria. The contract was sponsored by the Aero Propulsion Laboratory, Air Force Systems Command, Wright-Patterson AFB, Ohio 45433 under the direction of Sandra K. Drake (AFWAL/POTA), Project Engineer.



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## SECTION I

### INTRODUCTION

The material characterization tests discussed in this report are part of a foreign object damage (FOD) study of jet engine fan blades. The tests constitute a portion of the gross structural damage subtask of that study. The FOD study is designed to provide two resources necessary for the successful development of FOD resistant fan blades: (1) an analysis capability for predicting jet engine response to FOD impacts and (2) an understanding of the mechanisms of FOD failures. The fan blade analytical model uses inputs such as mechanical property data (of the material under consideration) and impact loads (representative of ice balls, ice spears, or small birds) to predict the fan blade response to the foreign object damage impacts. The model allows considerable evaluation of the blade design and material prior to the manufacturing and testing of a full stage of blades. This eliminates many costly experiments that have been necessary in the past.

The material characterization tests provide mechanical properties for use in the blade model discussed above. The materials tested include two metals, 410 stainless steel and 8Al-1Mo-1V titanium (used in the J-79 blade and F-101 blade, respectively) and two composites, boron/2024 aluminum and stainless steel wire mesh (used in the hybrid composite APSI blade). Part A of this report contains the description of the material characterization tests on the metals. Part B contains the discussion of the material characterization tests on the composite materials.

The mechanical properties of metals usually exhibit some strain rate dependence. (1,2,3) In addition, experiments have shown that the FOD impacts load various sections of the blades at different rates. For these reasons, tensile tests were conducted on the metals at strain rates ranging from  $10^{-3}$  to

about  $10^3$  strain/second. The quasi-static and intermediate rate tests ( $10^{-3}$  to 1 strain/second), conducted with an MTS closed-loop hydraulic system, provide the following elastic and plastic region parameters: Young's modulus, Poisson's ratio (in the elastic region), 0.2 percent offset yield stress, ultimate stress, and ultimate strain. The dynamic tests (500 to 700 strain/second), conducted on a split-Hopkinson bar apparatus, provide only the plastic region parameters: yield stress, ultimate stress, and ultimate strain. The mechanical properties are summarized in this report as plots of the various parameters versus strain rate.

## SECTION II

### EXPERIMENTAL PROCEDURES

Tensile tests conducted at the strain rates shown in Table 1 provide the desired mechanical property data. This range of rates encompasses the range of strain rates measured at select locations on the full scale blade tests, 12-370 strain/second. (The full scale blade tests were conducted as part of the gross structural damage subtask IV-A but reported with similar tests in the Task VI report.) The quasi-static and intermediate rate tests ( $10^{-3}$  to 1 strain/second) were run on an MTS electrohydraulic closed-loop testing machine. They require different specimens and procedures than do the dynamic tests, which were run on the split-Hopkinson bar apparatus. Consequently, the two types of tests are described separately below.

TABLE 1  
STRAIN RATES OF METALLIC MATERIAL  
CHARACTERIZATION TESTS

Material	Strain Rates (strain/second)
8Al-1Mo-1V Titanium	$10^{-3}$ , $10^{-1}$ , 1, 500
403 Stainless Steel	$10^{-3}$ , $10^{-1}$ , 1, 700

#### 2.1 LOW AND INTERMEDIATE STRAIN RATE TESTS

The information obtained from the low and intermediate strain rate tests include two elastic parameters, Young's modulus and Poisson's ratio, and three plastic parameters, the yield stress, the ultimate stress, and the ultimate strain. The following paragraphs describe the specimens used and how they were instrumented and tested. The full true stress-true strain curves appear in Appendix A.

### 2.1.1 Test Specimen and Specimen Instrumentation

The specimens were made from bar stock which had been worked the same amount as the finished blades. The specimen is illustrated in Figure 1. Its dimensions were per ASTM standard E8-69.<sup>(4)</sup> Three specimens of each material were tested at each rate with one exception. Only one specimen of the titanium was tested at 1 strain/second.

Figure 2 shows the location of measurement instrumentation placed on the specimens. The extensometer supplied information for obtaining the entire stress-strain from test start to specimen failure. The high resistance foil strain gage placed in the longitudinal direction (the same direction as the applied load) provided high resolution measurements for the parameters in the elastic region and at yield. The strain gage in the transverse direction (to the applied load) supplied measurements for the calculation of Poisson's ratio. The transverse strain gages were applied to only one specimen of each material at each strain rate. The data reduction equations appear in the analysis section of this report.

### 2.1.2 Test Procedures and Test System

The following paragraphs contain a brief description of the test system and test procedures used for the low and intermediate strain rate tests.

#### 2.1.2.1 Test System

The low and intermediate strain rate tests were conducted using the University's MTS electrohydraulic closed-loop testing system. This closed-loop test machine can be programmed in load, strain, or displacement control. The machine can produce displacement rates that range from less than 0.05 in/min. to 400 in/min. The signal conditioner used with the strain gage was a Honeywell 218 Bridge Amplifier. Included in the test system is an X-Y recorder and a multichannel transient recorder. An additional two pen X-Y recorder was used for these tests.

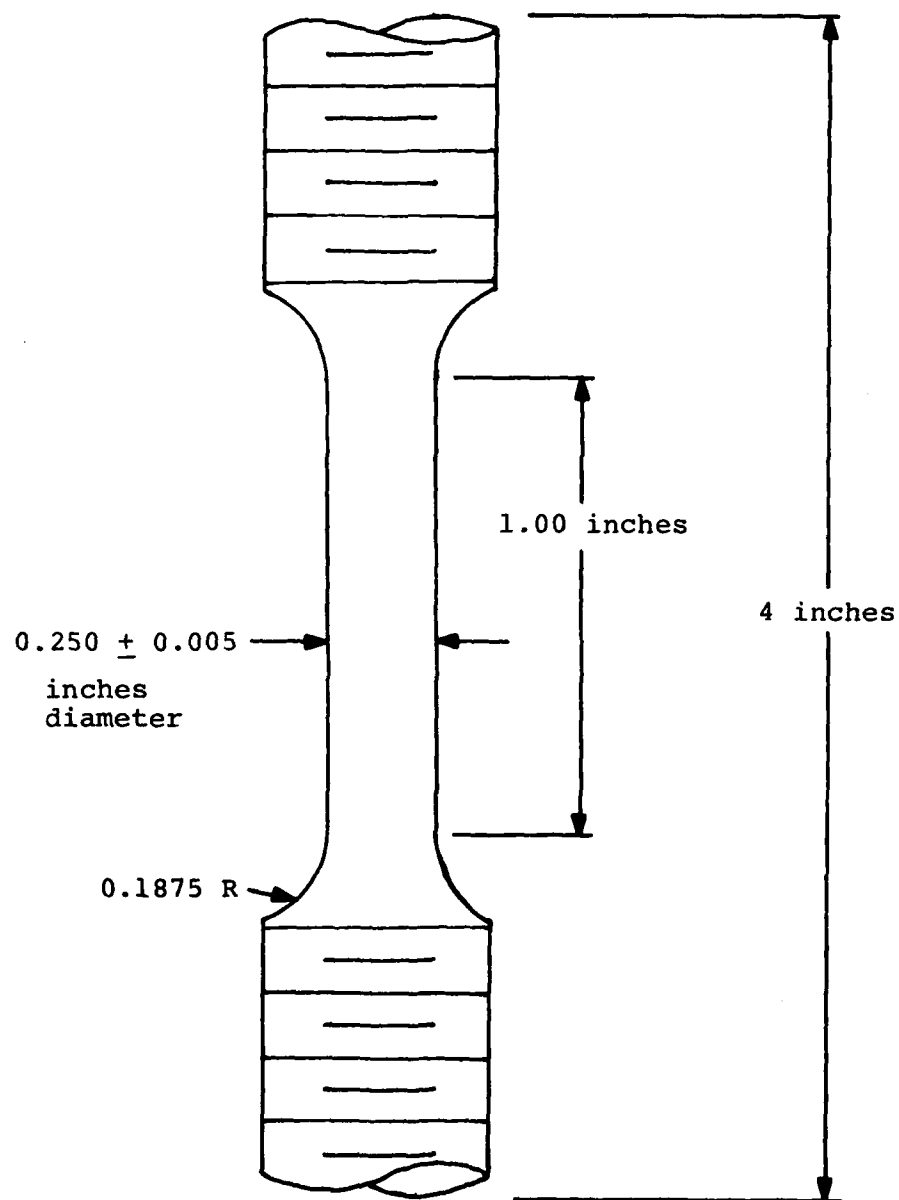
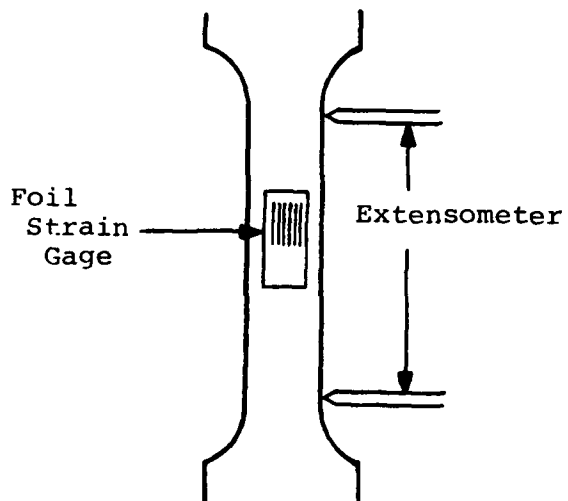
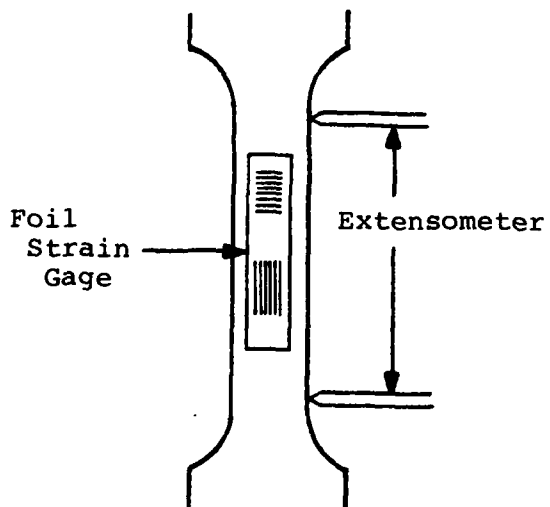


Figure 1. Specimen for Low and Intermediate Strain Rate Tests.



Specimen instrumented  
for axial strain and  
displacement  
measurement



Specimen instrumented  
for axial strain and  
displacement measure-  
ments and transverse  
strain measurements

Figure 2. Location of Measurement Instrumentation on the Low and Intermediate Strain Rate Specimens.

#### 2.1.2.2 Test Procedure

After tightening the strain gaged specimen in the grips, the extensometer was attached and adjusted to give the proper output voltage for the desired gage length. With the test machine in the load control mode a linear ramp function loaded the specimen to failure. Load and displacement were recorded directly on the system X-Y recorder for the lowest strain rate tests ( $10^{-3}$  strain/second). Load and strain were recorded directly on the additional X-Y recorder. For the higher strain rate tests ( $10^{-1}$  and 1 strain/second) the three parameters, load, strain, and displacement were recorded on three channels of the transient recorder for playback at slower speeds into the X-Y recorder.

### 2.2 HIGH STRAIN RATE TESTS

Mechanical properties obtained from the high strain rate tests include the following three plastic parameters: the yield stress, the ultimate stress, and the ultimate strain. Plastic region true stress-true strain curves were also acquired from the tests. (Elastic properties cannot be obtained from these tests because the pressure waves in the Hopkinson bar are not considered to be in equilibrium in the elastic region). The method (and apparatus) used is the same as that used in a study of the high strain rate tensile mechanical properties study of beryllium.<sup>(5)</sup> The subsequent paragraphs provide a detailed description of the experimental apparatus, the instrumentation, and the test procedures and mechanisms. The analysis section of this report contains the brief explanation of the data reduction process. The results appear in Appendix B.

#### 2.2.1 Experimental Apparatus

A schematic of the split Hopkinson bar apparatus and the associated instrumentation appears in Figure 3. Three parts of the apparatus are of major interest: two Hopkinson pressure bars (bar No. 1 and bar No. 2) and the striker bar.



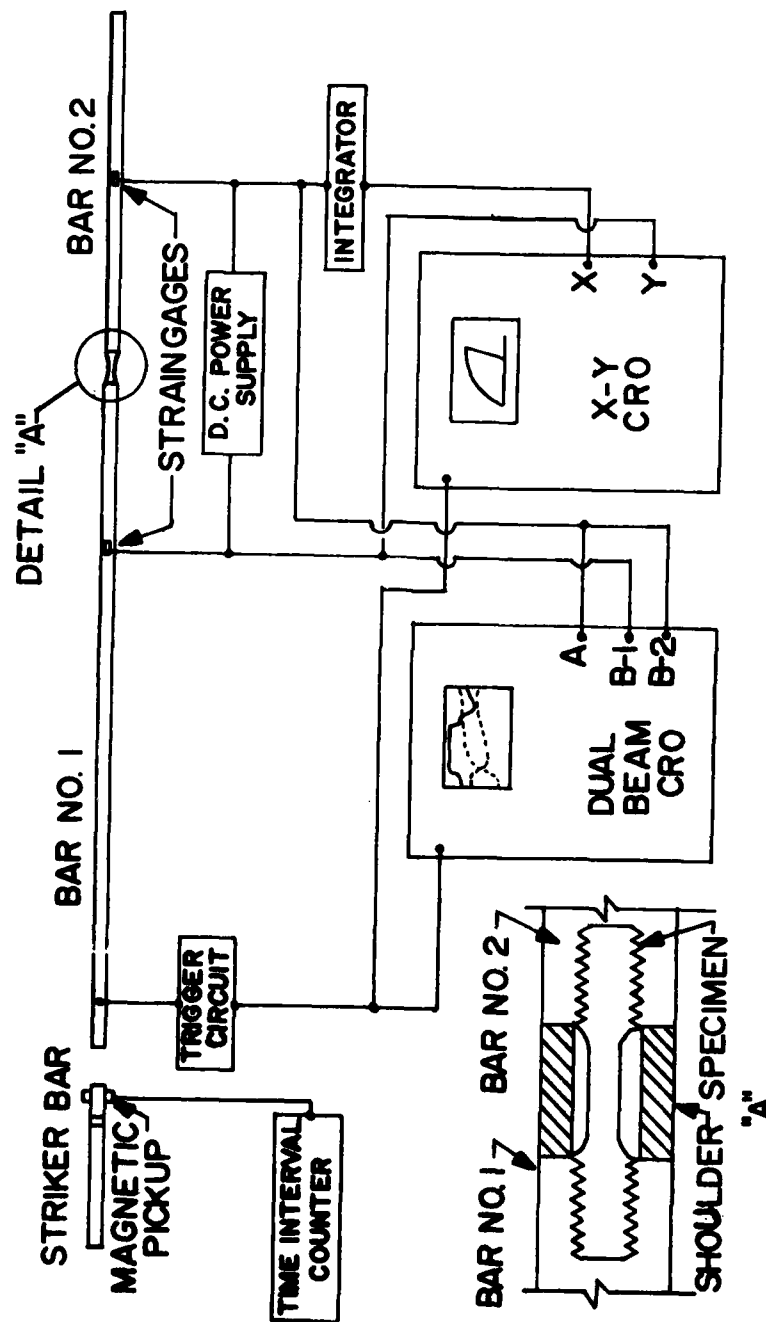


Figure 3. Schematic of Apparatus and Instrumentation for High Strain Rate Tests.

Bar No. 1 must be twice the length of bar No. 2 and the striker bar must be shorter than bar No. 2. For these experiments the bars are 0.06, 0.15, and 0.30 m lengths of 12.7 mm diameter AISI 4130 steel. They are mounted and aligned on a rigid base.

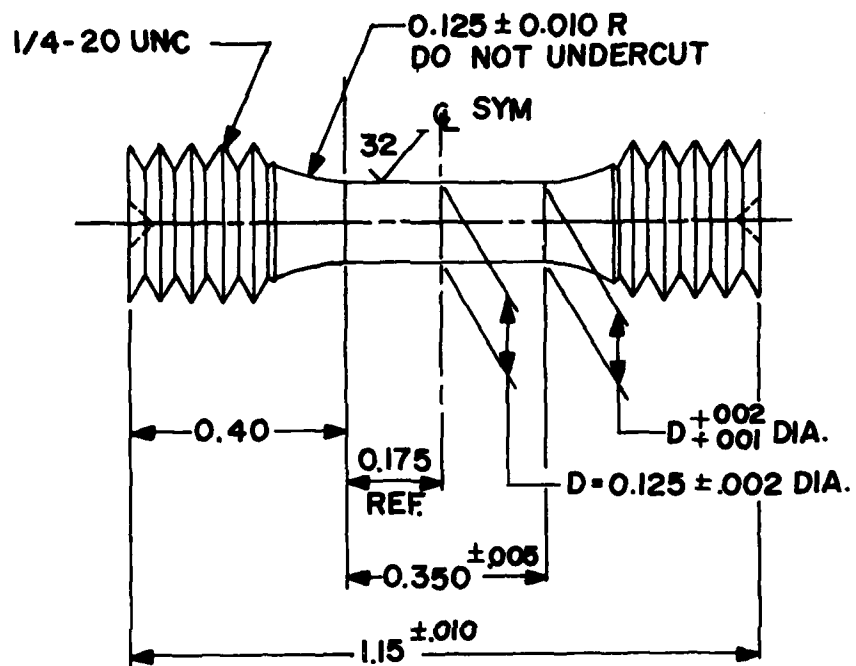
Detail "A" of Figure 3 shows an enlargement of a longitudinal section of the apparatus at the location of the specimen. To prepare for the test, the threaded tensile specimen, shown in Figure 4, is screwed partially into the pressure bars. Then a split shoulder is placed over the specimen, and the specimen is screwed in until the shoulder is snug against the pressure bars. The shoulder has the same outer diameter as the pressure bars (12.7 mm) and it has an inner diameter of 6.4 mm, just large enough to clear the specimen. The shoulder is made of the same material as the pressure bars, AISA 4130 steel. The ratio of the cross-sectional area of the shoulder to that of the pressure bar is 3:4, while the ratio of the area of the shoulder to the net cross-sectional area of the specimen is 12:1.

#### 2.2.2 Instrumentation

The recording circuitry consists primarily of strain gages, two oscilloscopes, and a counter. See Figure 3. The recording system is triggered when the striker bar impacts bar No. 1. A brief review of the major parts of the recording system follows.

Bars No. 1 and 2 are instrumented with high resistance foil strain gages. The gages on each bar are placed equidistant from the specimen so that the reflected and transmitted wave signals are time coincident. They are far enough from the specimen that no spurious reflections interfere with the pulses being recorded. Gages are placed diametrically opposite each other on the pressure bars to cancel bending.

The two oscilloscopes record data. One, a dual beam Tektronix type 565 oscilloscope, records the complete



NOTE: ALL DIMENSIONS IN INCHES

#### THREADED TENSION HOPKINSON BAR SPECIMEN

Figure 4. Specimen for High Strain Rate Tests.

strain time history of the test. One beam records the incident pulse as it passes the first gage. The second beam is used in chopped mode to record both the transmitted pulse (which is proportional to load) and the reflected pulse (which is proportional to strain rate). Figure 5(a) shows a photo of typical traces.

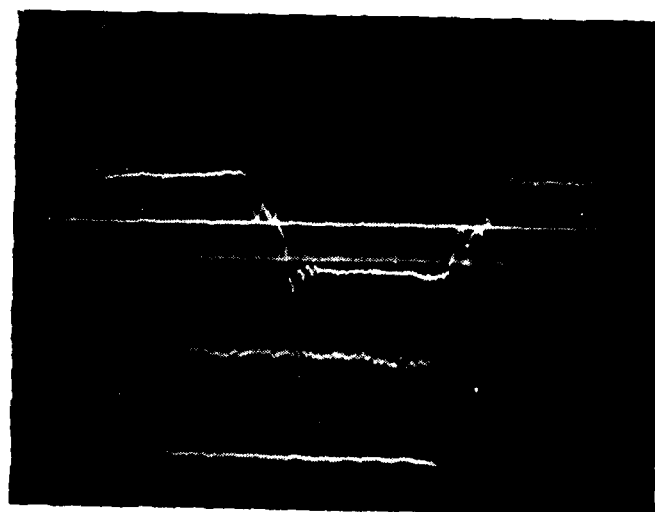
The second oscilloscope, an X-Y Tektronix, records a load-deflection curve. The reflected pulse (from gage No. 2) passes through an electronic integrator and is fed into the X axis of the scope. The signal is proportional to the displacement of the specimen. The signal from gage No. 1, proportional to the load in the specimen, is fed to the Y axis. Figure 5(b) shows the load-deflection trace.

### 2.2.3 Test Procedures and Mechanisms

The experiment begins when the striker bar is accelerated so that it impacts bar No. 1. The compression pulse generated travels down bar No. 1 to the specimen-shoulder junction. The pulse passes almost exclusively through the shoulder because of (1) the large area ratio of shoulder to specimen mentioned above and (2) the loose fit of the threaded joints (specimen and pressure bars). The portion of the pulse that passes through the specimen is below the elastic limit of the material. Then the compressive pulse travels through bar No. 2 to the free end, where it reflects as a tensile pulse. When this reflected pulse reaches the specimen, part of it is transmitted through the specimen and part is reflected back into bar No. 2 (because the shoulder doesn't carry the tensile pulse). The incident, transmitted and reflected pulses are recorded and analyzed for the required information.

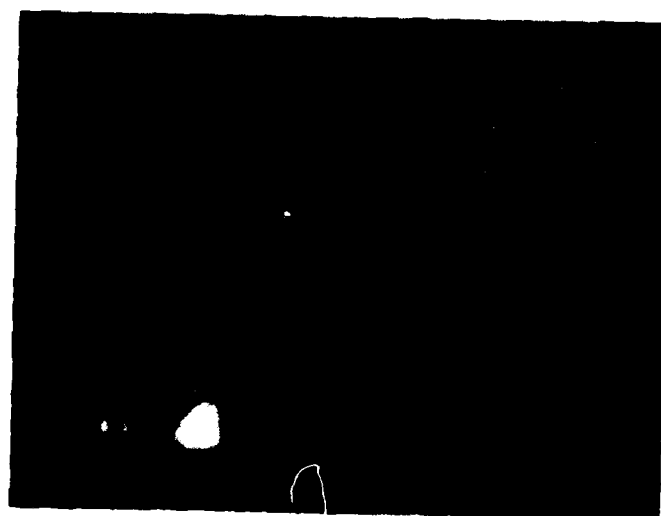
Figure 6 shows a Lagrangian x-t plot which illustrates the path of the pressure pulse during the test. The amplitude of the incident pressure pulse, generated by the striker bar impacting bar No. 1 is dependent on the striker bar velocity. Its length is twice the longitudinal elastic wave

$E_i$  —  
 $E_r$  —  
 $E_t$  —



a

Specimen: Ti-3



b

Vert: 465 lb/div.  
Horiz:  
0.00155 in/div.

Figure 5. Typical Oscilloscope Traces.

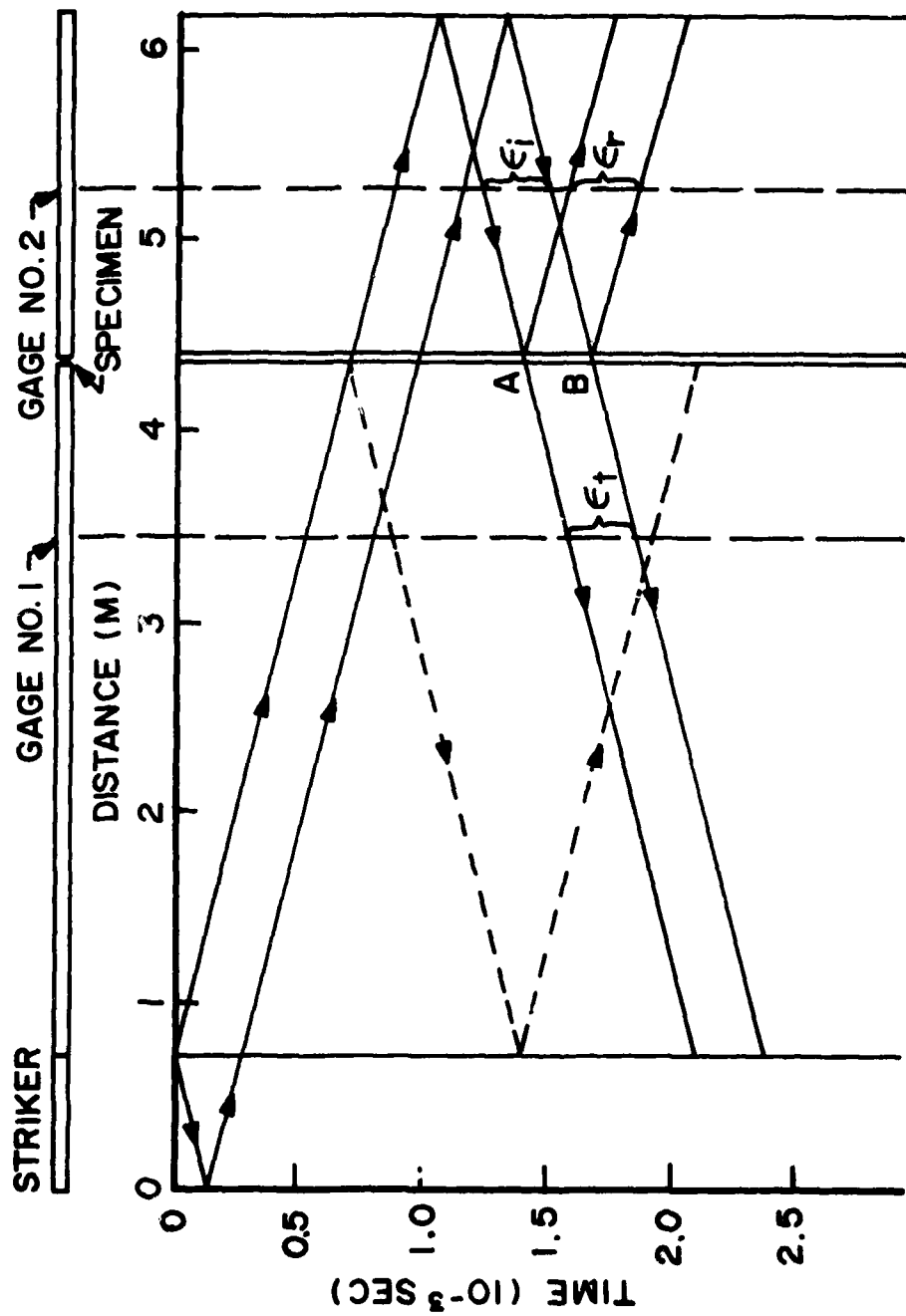


Figure 6. Lagrangian Displacement - Time Plot for High Strain Rate Tests.

transit time in the striker bar. The plot shows that the compressive pulse passes through the entire apparatus and is reflected from the free end of bar No. 2 as a tensile pulse,  $\epsilon_i$ . The plot also illustrates the tensile pulse splitting when it reaches the specimen-shoulder junction and part of it,  $\epsilon_t$ , being transmitted through the specimen while part,  $\epsilon_r$ , is reflected back into bar No. 2.

### SECTION III ANALYSIS

This section outlines the data reduction techniques and equations used to obtain the desired true stress-true strain curves and specific elastic and plastic parameters. All of the test data began as analog load-displacement curves that were then analyzed by various computer routines particular to the test type (low and intermediate, or high strain rate). The results appear in the appendices in analog and digital form.

#### 3.1 LOW AND INTERMEDIATE STRAIN RATE TESTS

The low and intermediate strain rate tests resulted in (1) load-displacement curves from the extensometer and (2) load-strain curves from the strain gages. Converting the displacement measurement to true strain is relatively simple. Using  $u$  as displacement and  $l_0$  as the initial gage length, one obtains the following:

$$\epsilon_E = \frac{u}{l_0} \quad (\text{engineering strain}) \quad (1)$$

$$\epsilon_T = \ln (1 + \epsilon_E) \quad (\text{true strain}) \quad (2)$$

where  $\ln$  is the natural logarithm.

The extensometer gage length,  $l_0$ , was 1 inch for these tests. Consequently, the digitized displacement value could be used directly in calculations as engineering strain.

The strain gage data was also simple to reduce. A correction factor was applied to the strain reading to account for the nonlinearity of the bridge circuit. The incremental error,  $n$ , is added to the indicated,  $\hat{\epsilon}$ , to obtain the actual strain,  $\epsilon$ , causing the resistance change in the gage. For this case, a single active gage in a quarter-bridge arrangement, the



correction is

$$n = \frac{F (\hat{\epsilon})^2 \times 10^{-6}}{2 - F (\hat{\epsilon}) \times 10^{-6}} \quad (3)$$

where  $F$  is the gage factor. Equation 2 is then used to obtain true strain.

Obtaining true stress values for the low and intermediate strain rate tests is more complicated. True stress,  $\sigma_T$ , is proportional to engineering stress,  $\sigma_E$ , by the ratio of the initial area,  $A_0$ , to the current area,  $A$ . In the elastic region, where one has small strains, the ratio of areas can be expressed in terms of the longitudinal engineering strain so that

$$\sigma_T = \frac{\sigma_E (1 + \epsilon_E)}{1 + \epsilon_E (1 - 2\nu)} \quad \begin{matrix} \text{(elastic region} \\ \text{true stress)} \end{matrix} \quad (4)$$

In the plastic region one assumes the material is incompressible, consequently the volume remains constant. Based on this assumption the true stress in the plastic region becomes

$$\sigma_T = \frac{\sigma_E (1 + \epsilon_E)}{1 + \bar{\epsilon}_E (1 - 2\nu)} \quad \begin{matrix} \text{(plastic region} \\ \text{true stress)} \end{matrix} \quad (5)$$

where  $\bar{\epsilon}_E$  is the engineering strain evaluated at the yield point of the material.

Three elastic region parameters were calculated from the strain gage data. Young's modulus is the slope of the linear part of the curve. Poisson's ratio,  $\nu$ , was obtained for one test per material per strain rate with the following relationship:

$$\nu = +\epsilon_{\text{transverse}} / \epsilon_{\text{longitudinal}}$$

The shear modulus,  $G$ , was calculated with the equation

$$G = \frac{E}{2(1 + \nu)}$$

The values of these mechanical properties appear in the results section. An additional parameter, the yield stress, was obtained from these curves. It was evaluated using the 0.2 percent (strain) offset method because neither of the metals exhibits a definite yield point. The load-displacement curves were analyzed using Equations 2, 4, and 5 to produce the full true stress-time strain curves. The ultimate strain was taken as the final strain value of the curve. However, the ultimate stress was calculated using the final load divided by the final reduced area of the specimens because Equation 5 is not accurate (complete) for large amounts of plasticity.

These curves were all digitized on the University of Dayton's Tektronix 4014-1 computer graphics terminal and associated digitizing board.

### 3.2 HIGH STRAIN RATE TESTS

The polaroid photographs of the load-deflection traces obtained from the high strain rate tests were digitized using a Hewlett Packard 9800 series calculator and digitizer (located at the AFML) to produce true stress-engineering strain curves. Those curves were then digitized on the Tektronix 4014-1 computer graphics terminal and digitizing board (located at the University) to produce true stress-true strain curves. A brief review of the analysis in the two data reduction programs follows. Details of the theory of the measurements appear in documents by U. S. Lindholm<sup>(6)</sup> and by Lindholm and L. M. Yeakley.<sup>(7)</sup>

The average stress in the specimen is:

$$\sigma_s = E (A/A_s) \epsilon_t \quad (6)$$

where E and A are the elastic modulus and cross-sectional area of the pressure bars,  $A_s$  is the cross-sectional area of the specimen and  $\epsilon_t$  is the transmitted wave. Using the assumption of no volume change after first yield, Equation (6) is modified to:

$$\sigma_T = E(A/A_s) \epsilon_t \frac{1}{(1 + \epsilon_t)} \quad (7)$$

for true stress.

Engineering strain was obtained from the deflection values using the following relationship:

$$\epsilon = 0.265 \delta - 0.5 (1 - \exp^{-0.55 \delta})$$

where  $\epsilon$  represents strain in percent and  $\delta$  represents the deflection in mils between the grips.<sup>(5)</sup> This equation results from least-squares fitting many deflection-strain plots from several different materials. It is particular to the specimen size used, not to a material.

The yield stress was determined as the point on the stress-strain curve at which the increase in applied load essentially ceased. The ultimate stress and ultimate strain were selected as the final point on the stress-strain curve. Reasons for this selection procedure appear in the results and discussion section.

## SECTION IV

### RESULTS AND DISCUSSION

The mechanical properties acquired from the various strain rate tests on 8Al-1Mo-1V titanium and 410 stainless steel appear in Tables 2 and 3, respectively. Table 4 shows the measured density of each material. Figures 7 through 10 are plots of the mechanical properties versus strain rate. The points lacking standard deviation marks either represent one data point or they represent averaged data having a standard deviation smaller than the data symbol. The true stress-true strain curves for the low and intermediate tests (Appendix A) have two parts to the curve in the plastic region. The solid line represents the test data analyzed with the plastic region equations given in the analysis section. Obviously these are not complete for large strains. The dotted line joins the ultimate stress (calculated with the final load and reduced cross-sectional area)-ultimate strain point to the test curve.

Studying Figures 7 through 10, one concludes that all of the tensile mechanical properties determined by these tests vary with strain rate with the exception of the elastic modulus of titanium. Perhaps additional testing of the stainless steel would reveal that its elastic modulus also does not vary with strain rate. It is important to consider the comments in the following paragraphs when using the curves shown as data for calculations.

A very small specimen must be used for high strain rate tests to satisfy assumptions made in the Hopkinson bar theory equations. This presents problems in obtaining accurate data from the tests, particularly in the elastic region. One problem is that achievable machining tolerances on the specimen in relation to load surface alignment make it difficult to resolve small strains accurately. Secondly, an assumption made in developing the Hopkinson bar equations is that many stress

TABLE 2  
MECHANICAL PROPERTIES OBTAINED FOR 8Al-1Mo-1V TITANIUM

Specimen	Strain Rate ( $\epsilon/\text{sec}$ )	Elastic Modulus ( $\times 10^6$ psi)	Shear Modulus ( $\times 10^6$ psi)	Yield Stress (ksi)	Ultimate Stress (ksi)	Ultimate Strain (percent)
1-Ti	0.001	16.9	6.5	148.9	202.5	28.2
2-Ti	0.001	17.3	6.7	148.7	200.8	27.2
3-Ti	0.001	17.4	6.8	148.7	199.3	21.8
4-Ti	0.10	15.8	6.3	156.3	188.5	13.7
5-Ti	0.10	16.9	6.8	153.0	189.1	15.9
6-Ti	0.10	17.1	6.8	156.4	194.5	15.3
7-Ti	1.0	17.3	6.8	164.0	184.6	15.7
Ti-3	550	*	*	203.0	161.1	13.7
Ti-2	560	*	*	202.0	154.2	14.5
Ti-1	580	*	*	208.0	158.7	15.3

Poisson's Ratio = 0.279

\* Elastic properties are not obtainable from the high strain rate tests.

TABLE 3  
MECHANICAL PROPERTIES OBTAINED FOR 410 STAINLESS STEEL

Specimen	Strain Rate ( $\epsilon/\text{sec}$ )	Elastic Modulus ( $\times 10^6$ psi)	Shear Modulus ( $\times 10^6$ psi)	Yield Stress (ksi)	Ultimate Stress (ksi)	Ultimate Strain (percent)
4-SS	0.001	27.4	10.6	84.2	207.2	26.7
5-SS	0.001	30.5	11.7	84.8	207.2	27.2
6-SS	0.001	33.7	12.9	84.9	211.9	26.9
9-SS	0.10	27.7	11.2	87.2	191.9	25.7
10-SS	0.10	26.4	10.2	87.0	188.3	26.8
11-SS	0.10	28.2	10.9	89.2	198.3	23.5
12-SS	1.0	27.2	10.4	86.6	182.7	23.8
13-SS	1.0	26.1	10.0	91.3	185.5	25.4
14-SS	1.0	27.2	10.4	94.5	183.8	24.6
15-SS	1.0	28.0	10.9	88.8	184.7	25.6
SS-5	680.0	*	*	124.0	129.5	16.9
SS-4	760.0	*	*	123.0	130.0	17.2
SS-3	700.0	*	*	131.0	131.4	18.0
SS-2	530.0	*	*	128.0	**	**
SS-1	340.0	*	*	126.0	**	**

Poisson's Ratio = 0.297

\* Elastic properties are not obtainable from the high strain rate tests.

\*\* These specimens did not fail.

TABLE 4  
DENSITY

Material	Density
8Al-1Mo-1V Titanium	4.32 g/cm <sup>3</sup>
410 Stainless Steel	7.57 gm/cm <sup>3</sup>

wave reflections occur in the specimen before a state of dynamic equilibrium exists. This does not happen until some plastic deformation has taken place. In addition, once necking occurs in the specimen a uniform stress field no longer exists making the equations for stress and strain invalid. This is a significant factor for the small specimens.

However, the split-Hopkinson bar apparatus represents the state-of-the-art technique for obtaining stress-strain information at high strain rates. The curves are considered most accurate in the range of 2 to 10 percent strain. The yield stress values reported represent the point on the curves where the load ceases to increase significantly for a rather large increase in displacement. (The 0.2 percent offset method obviously cannot be used because of the lack of data in the elastic region.) The ultimate stress and ultimate strain values from these tests were the final point of the curve. Although the values from the high strain rate tests cannot be used as exact data points they are useful in establishing the trend of the strain rate dependence of the subject materials over nearly seven decades of strain rate.

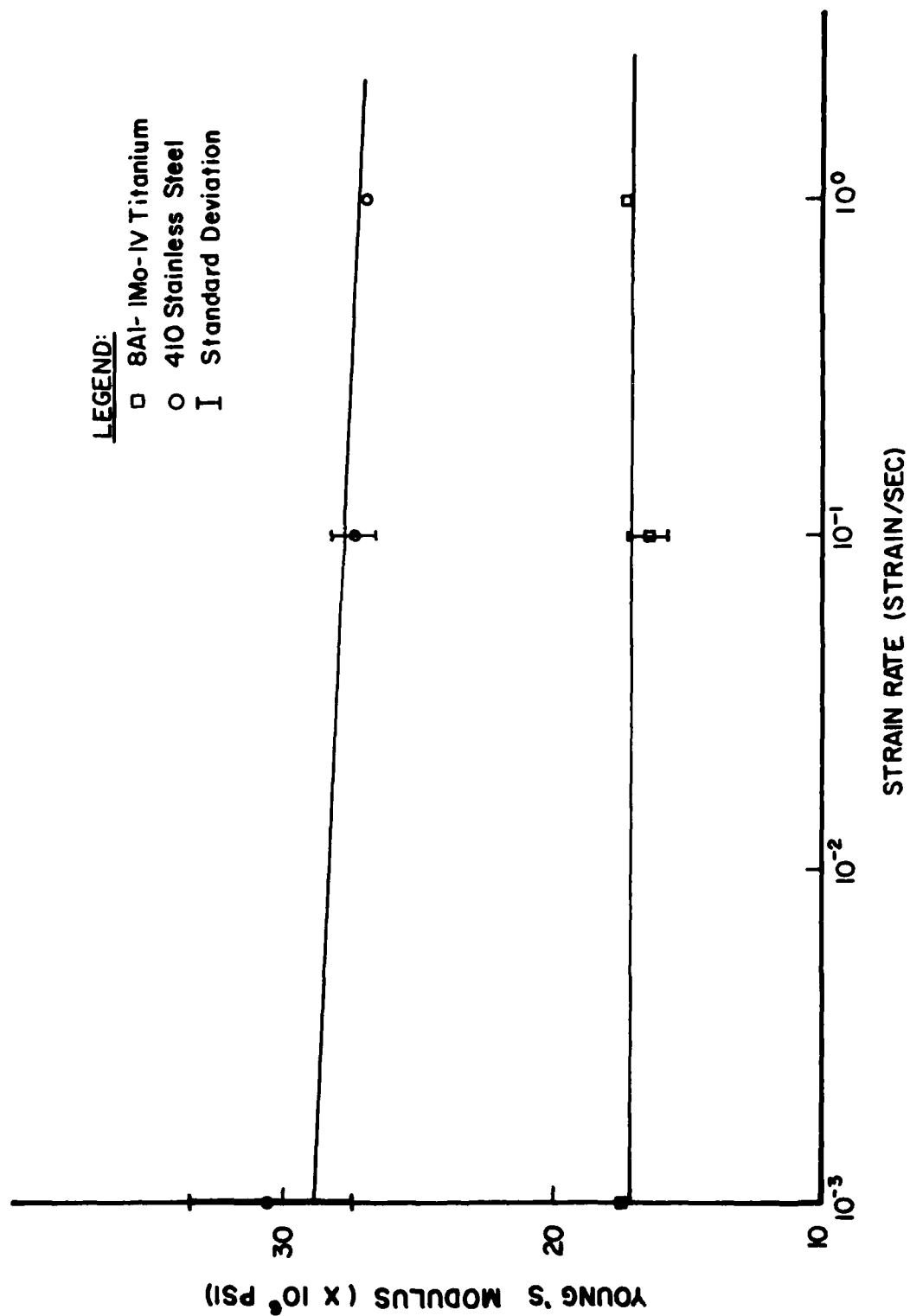


Figure 7. Young's Modulus VS. Strain Rate.



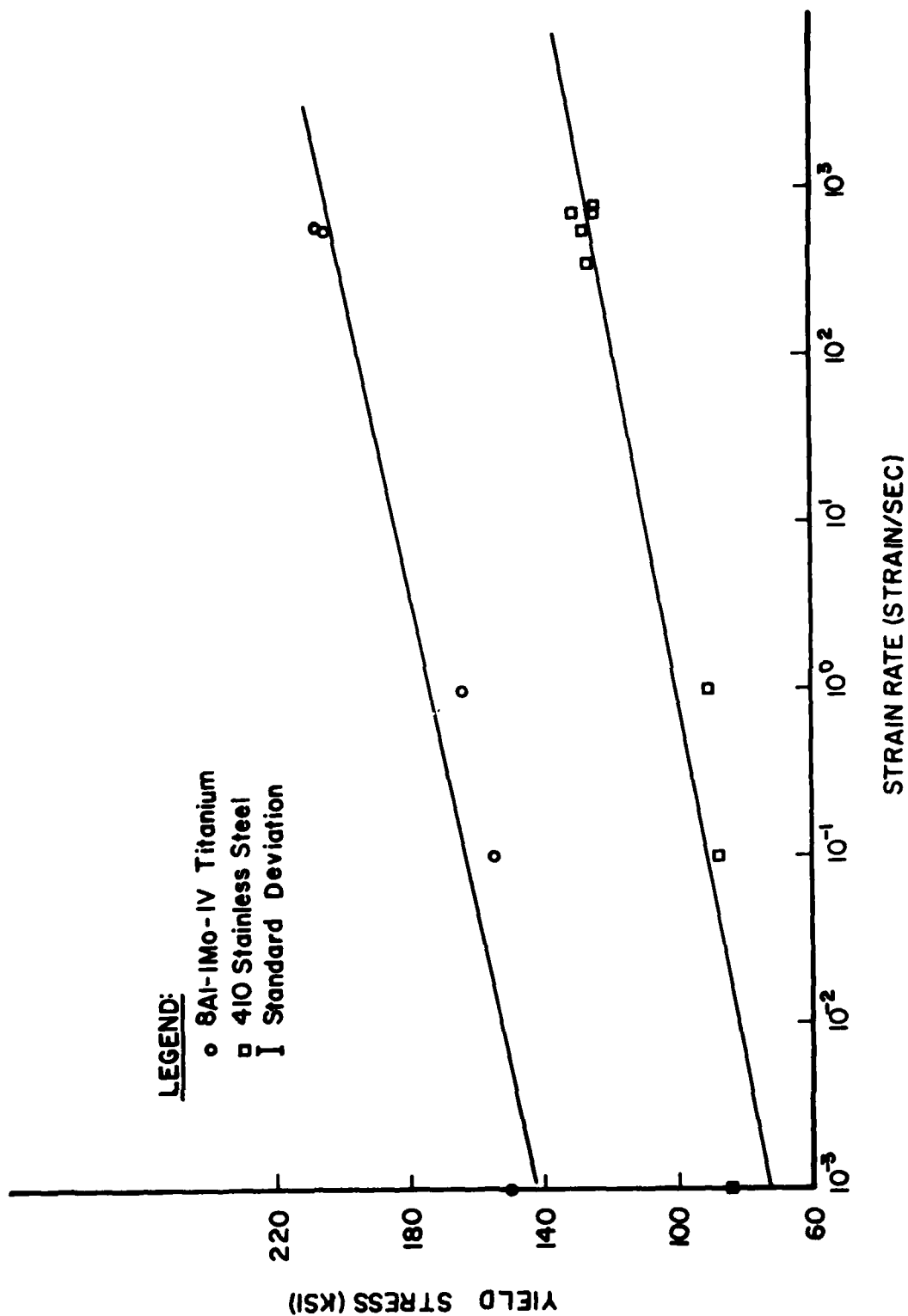


Figure 8. Yield Stress VS. Strain Rate

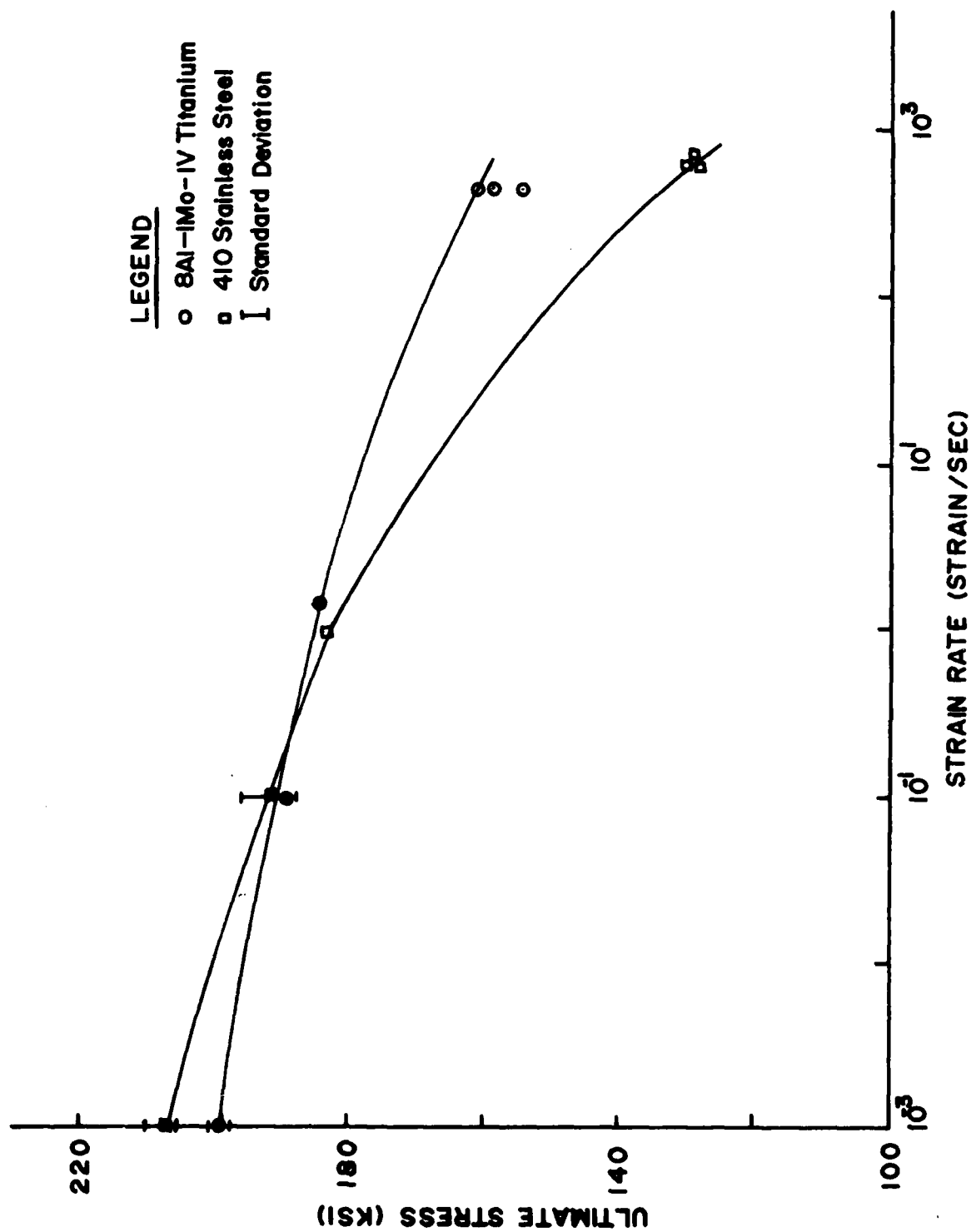


Figure 9. Ultimate Stress VS. Strain Rate.

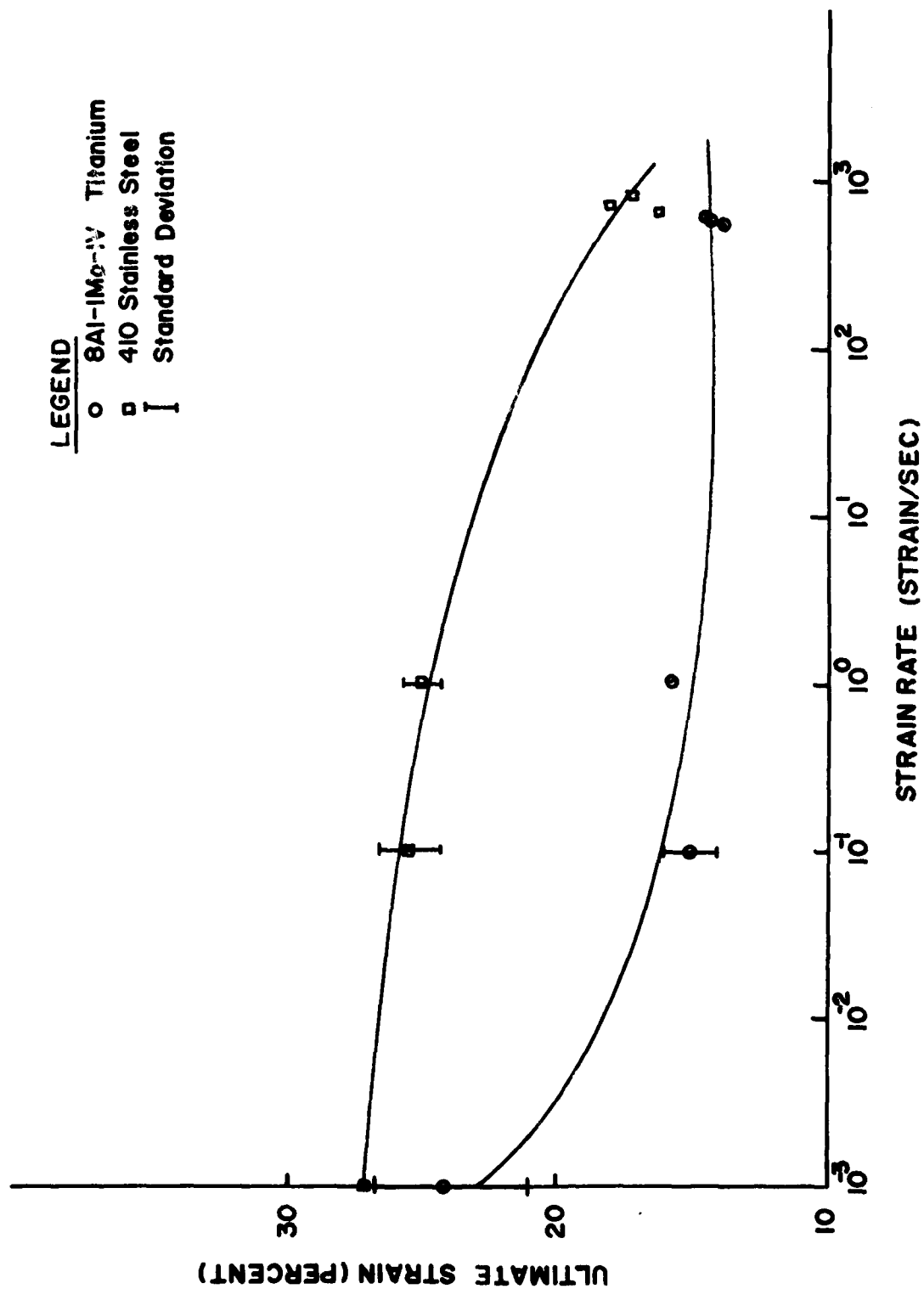


Figure 10. Ultimate Strain VS. Strain Rate.

## REFERENCES

1. Mardirosian, M.M., "Strain Rate Effects in Brittle and Tough Materials," AMMRC-TR-74-34, December 1974.
2. Green, S.J. and S.G. Babcock, "Response of Materials to Suddenly Applied Stress Loads: Part I: High Strain-rate Properties of Eleven Reentry-vehicle Materials at Elevated Temperatures," TR66-83 Part I, November 1966.
3. Eleiche, Abdel-Salam M., "A Literature Survey of the Combined Effects of Strain Rate and Elevated Temperature on the Mechanical Properties of Metals," AMFL-TR-72-125, September 1972.
4. 1977 Annual Book of American Society for Testing and Materials Standards, Part 10: Metals Physical, Mechanical, Corrosion Testing, pp. 154-173.
5. Nicholas, T., "Mechanical Properties of Structural Grades of Beryllium at High Strain Rates," AFML-TR-75-168, October 1975.
6. Lindholm, U.S., "Some Experiments with the Split Hopkinson Pressure Bar," Journal of the Mechanics and Physics of Solids, 1964, Vol. 12, pp. 317-335.
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APPENDIX A

LOW AND INTERMEDIATE STRAIN RATE  
CURVES AND DIGITAL DATA

STANDARD TENSILE			STANDARD TENSILE		
0.001 STRAIN/SEC			0.001 STRAIN/SEC		
BAL-1M0-1U TITANIUM			BAL-1M0-1U TITANIUM		
SPEC 1-T1			SPEC 2-T1		
1	0.5808615	7.606	1	0.3439573	4.748
2	1.3095612	18.700	2	0.6333684	10.304
3	1.9177013	29.176	3	0.13626456	20.631
4	2.5369049	38.436	4	0.19605711	30.739
5	3.1774354	49.340	5	0.26735502	42.304
6	3.9414203	61.289	6	0.36057480	58.211
7	4.5869500	72.210	7	0.46522603	66.695
8	5.1532190	82.512	8	0.52186549	75.591
9	5.8442455	92.229	9	0.58173847	84.093
10	6.6005061	105.440	10	0.6323282	93.835
11	7.2509252	114.962	11	0.68336169	101.921
12	7.9313489	126.747	12	0.73925781	109.409
13	8.6556287	137.730	13	0.81140866	118.749
14	9.2657194	146.856	14	0.87203732	128.750
15	1.00219249	150.487	15	0.92116468	137.296
16	1.11203352	150.089	16	0.97227816	144.385
17	1.24609528	149.756	17	1.03806357	148.820
18	1.39599318	149.741	18	1.10983378	150.428
19	1.60598576	150.503	19	1.20642624	149.061
20	1.79960876	150.811	20	1.36910988	149.394
21	1.98463584	151.511	21	1.52336818	149.495
22	2.16923089	152.419	22	1.70183256	150.296
23	2.31751753	153.025	23	1.8857076	150.713
24	2.51358850	153.762	24	2.20703259	151.784
25	2.69730043	154.468	25	2.37588535	152.769
26	2.91653145	155.274	26	2.54064246	153.328
27	3.2579162	156.814	27	2.70497600	154.006
28	3.5478703	157.218	28	2.92250511	154.840
29	3.7094237	157.667	29	3.19404027	155.098
30	3.8599825	158.083	30	3.4766243	156.023
31	4.03346833	158.567	31	3.5612205	156.127
32	4.21056616	159.062	32	3.72316532	156.817
33	4.4237782	159.500	33	3.8571937	157.188
34	4.63441857	160.041	34	4.0425547	157.755
35	4.81054619	160.325	35	4.17390316	157.887
36	4.9636415	160.610	36	4.35982825	158.342
37	5.16559302	161.332	37	4.49355161	159.208
38	5.34476368	161.416	38	4.65903757	159.244
39	5.51047106	162.129	39	4.79191517	159.417
40	5.6437597	162.000	40		159.882
41	5.76375881	162.615			
42	5.86507948	164.912			
43	19.06724825	186.435			
44	19.85094865	188.399			
45	20.42284516	189.761			
46	20.80217157	190.485			
47	21.35157297	191.457			
48	21.89759980	191.733			
49	22.35680507	192.485			
50	22.81458814	192.326			
51	23.5098873				

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13. 579375274		162.140		25. 2071079176		177.843	
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22. 216870509		178.270		34. 05214512		15.998	
23. 2145459210		175.166		35. 06627623		19.771	
24. 2166894962		172.637		36. 08049998		23.340	
25. 2185567641		170.057		37. 10195010		28.441	
26. 2205201694		167.465		38. 11238959		31.603	
27. 2218952346		164.712		39. 11876524		34.967	
				40. 13740770		41.093	
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				42. 1876524		49.365	
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				45. 20780840		59.982	
				46. 22616511		65.310	
				47. 25937818		68.899	
				48. 28942161		71.974	
				49. 3117721		75.571	
				50. 34971736		78.768	
				51. 39575265		81.255	
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				53. 51209110		85.887	
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				56. 85664452		91.336	
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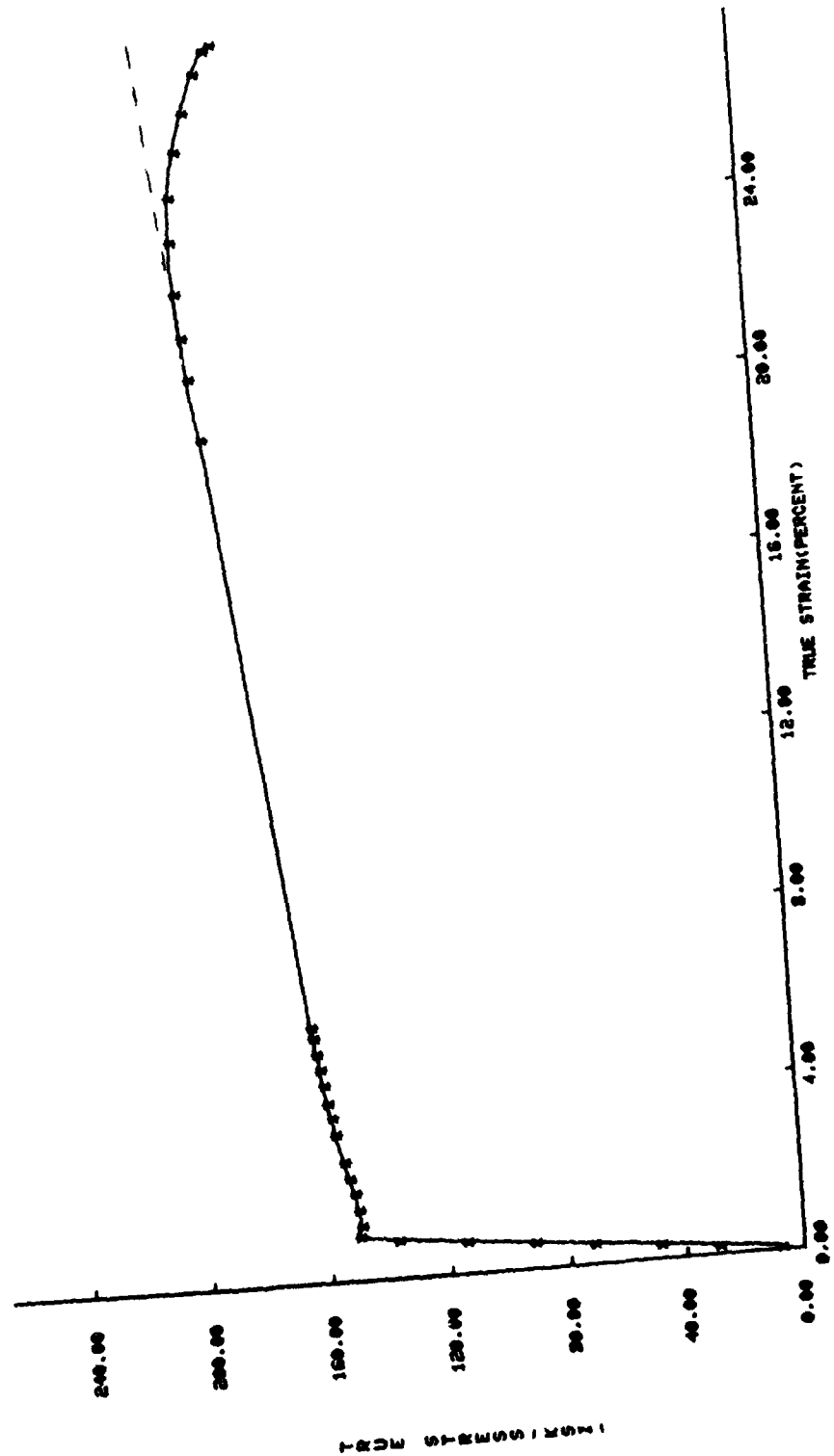
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79	26.53849205	75.504	51	22.59968014	114.242
80	26.70388300	72.860	52	23.07143095	112.461
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82	26.90233717	69.436	54	23.87460523	108.692
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			57	24.9350716	100.728
			58	25.62243481	94.388
			59	25.89944867	91.215
			60	26.12703423	88.642
			61	26.32146874	86.030
			62	26.53158584	83.022
			63	26.75751277	80.013
			64	27.0880156	74.660
			65	27.19091102	72.605
			66	27.31920534	70.823
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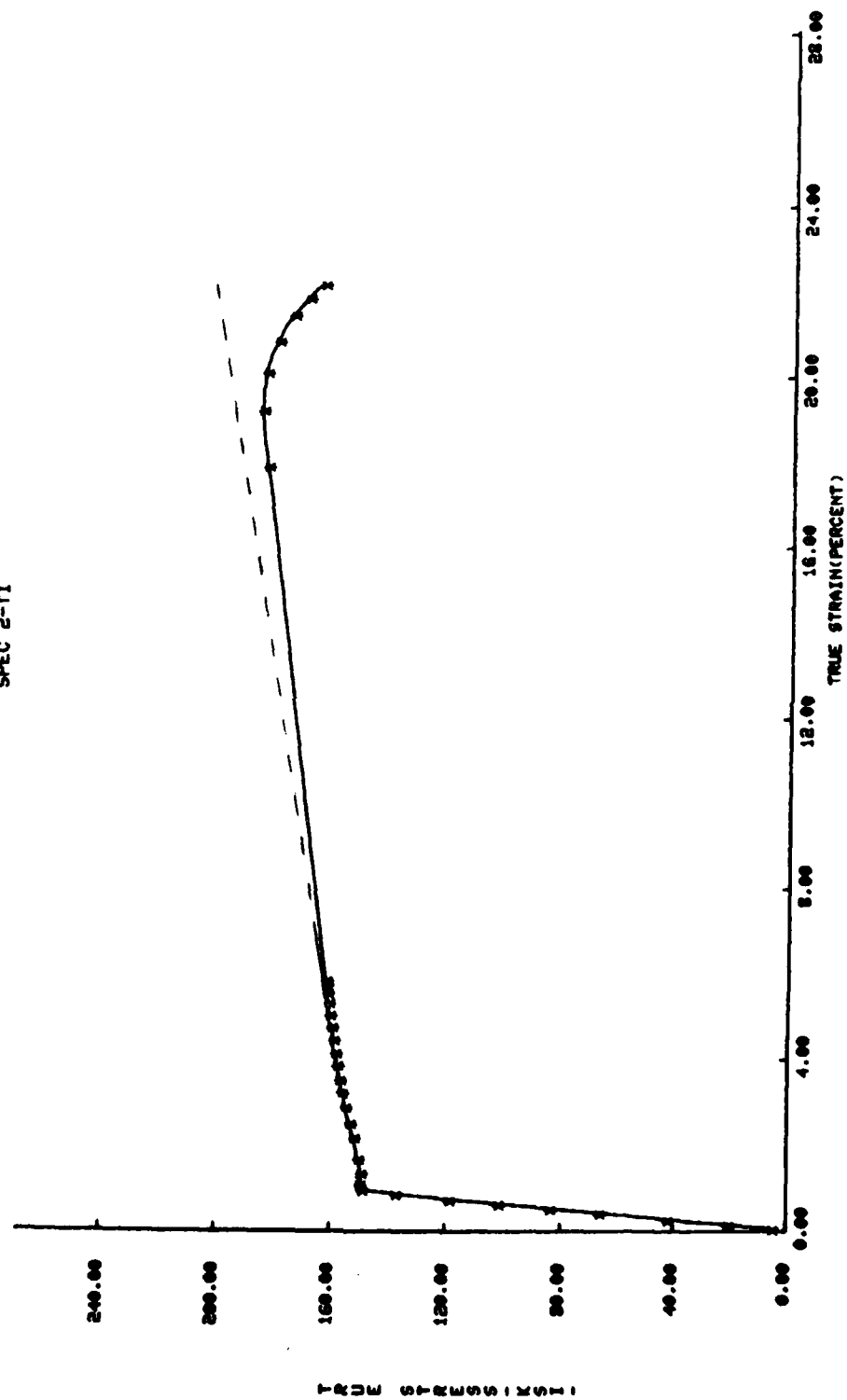
410 STAINLESS STEEL  
SPEC 6-SS

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4	.02093985	10.885	60	25.25290308	95.005
5	.02675336	14.854	61	25.51848323	90.331
6	.04102432	19.941	62	25.88805782	86.596
7	.05491508	24.214	63	26.09287394	83.859
8	.08136758	29.912	64	26.36097701	80.356
9	.10441304	36.835	65	26.59669600	77.077
10	.11867382	41.926	66	26.75207241	73.863
11	.1412820	48.240	67	26.87562877	70.886
12	.16790905	54.047			
13	.19017616	59.346			
14	.21201755	63.729			
15	.24583095	68.318			
16	.28349775	72.705			
17	.31657978	75.768			
18	.38972589	79.961			
19	.46629866	83.138			
20	.54243852	85.513			
21	.62626980	87.603			
22	.70590070	89.284			
23	.73757569	89.610			
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25	1.01812953	92.237			
26	1.22289352	93.699			
27	1.48546666	94.172			
28	1.75914905	94.861			
29	1.99720336	95.841			
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31	2.39365903	96.719			
32	2.68037895	97.531			
33	2.86583389	97.861			
34	3.02398480	98.174			
35	3.24344844	98.632			
36	3.44311656	98.765			
37	3.66927254	99.125			
38	3.79931531	99.212			
39	3.97503224	99.434			
40	4.15048632	99.762			
41	4.27619180	100.058			
42	4.39414048	100.349			
43	4.49770933	100.486			
44	16.95228513	114.734			
45	17.53303671	115.354			
46	18.2269361	116.228			
47	18.87494346	116.678			
48	19.54729098	117.037			
49	20.31578132	117.003			
50	20.84261732	117.075			
51	21.48523133	116.890			
52	21.93722072	115.861			
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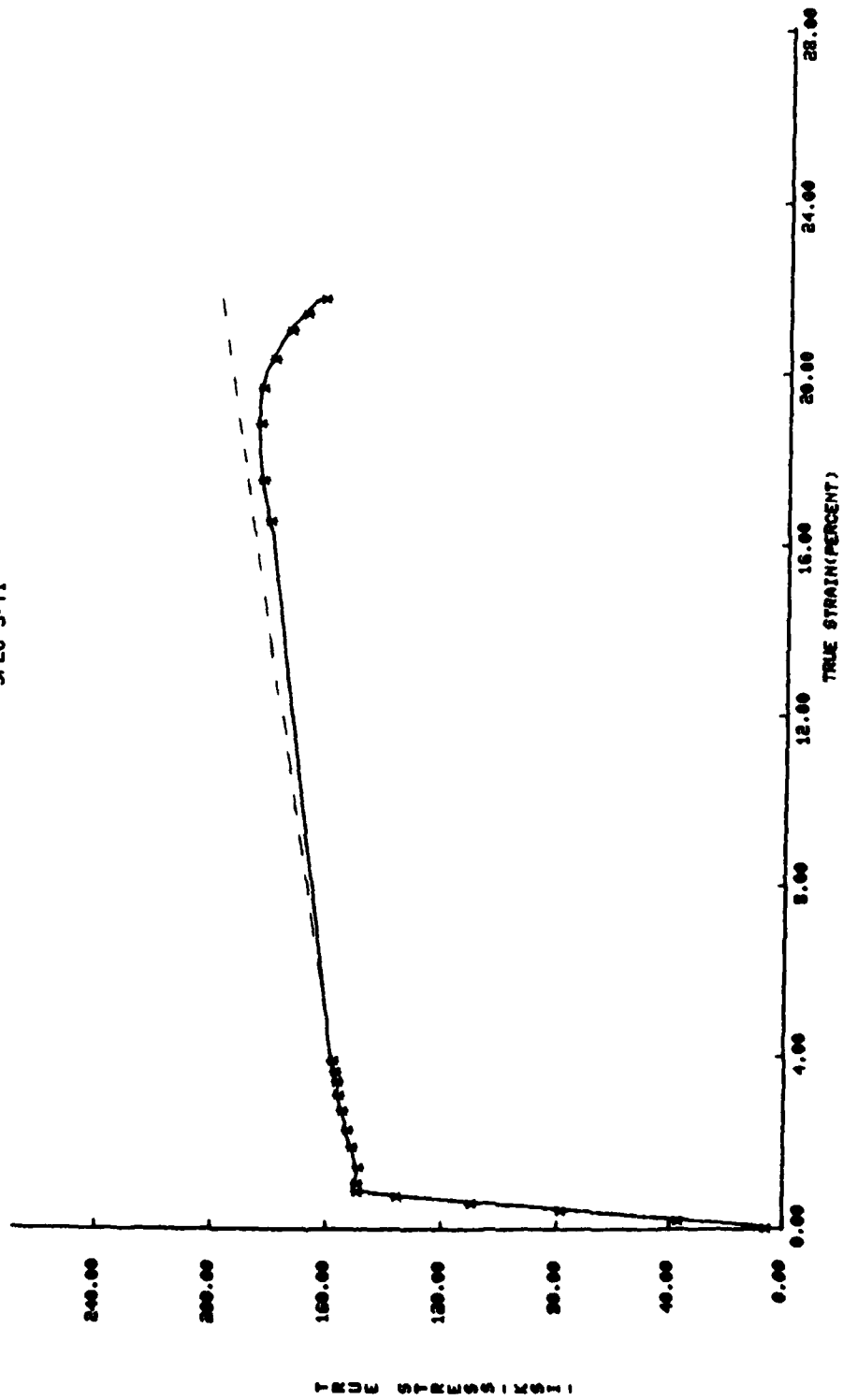
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BAL-100-10 TITANIUM  
SPEC 1-T1



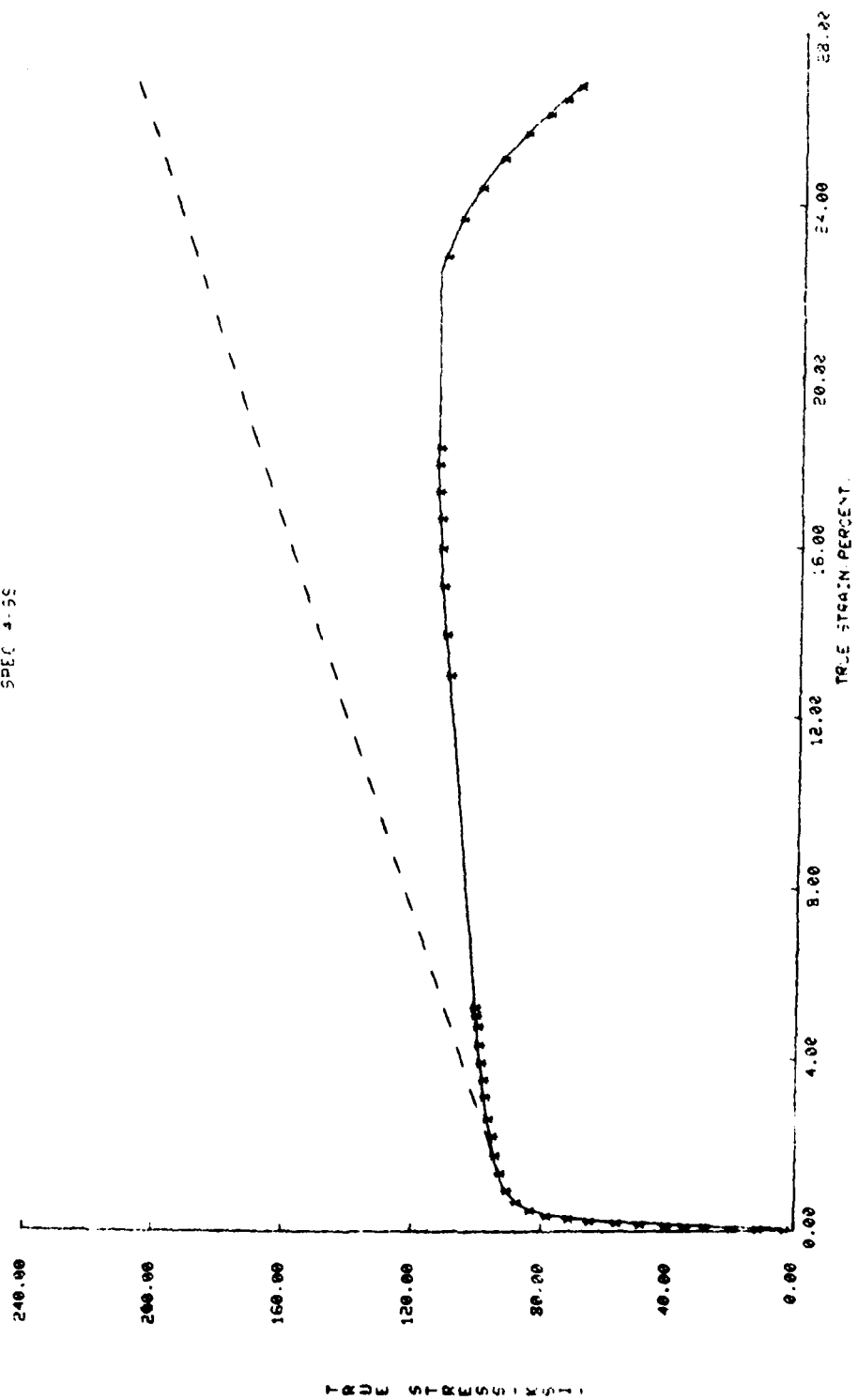
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SPEC 2-TI



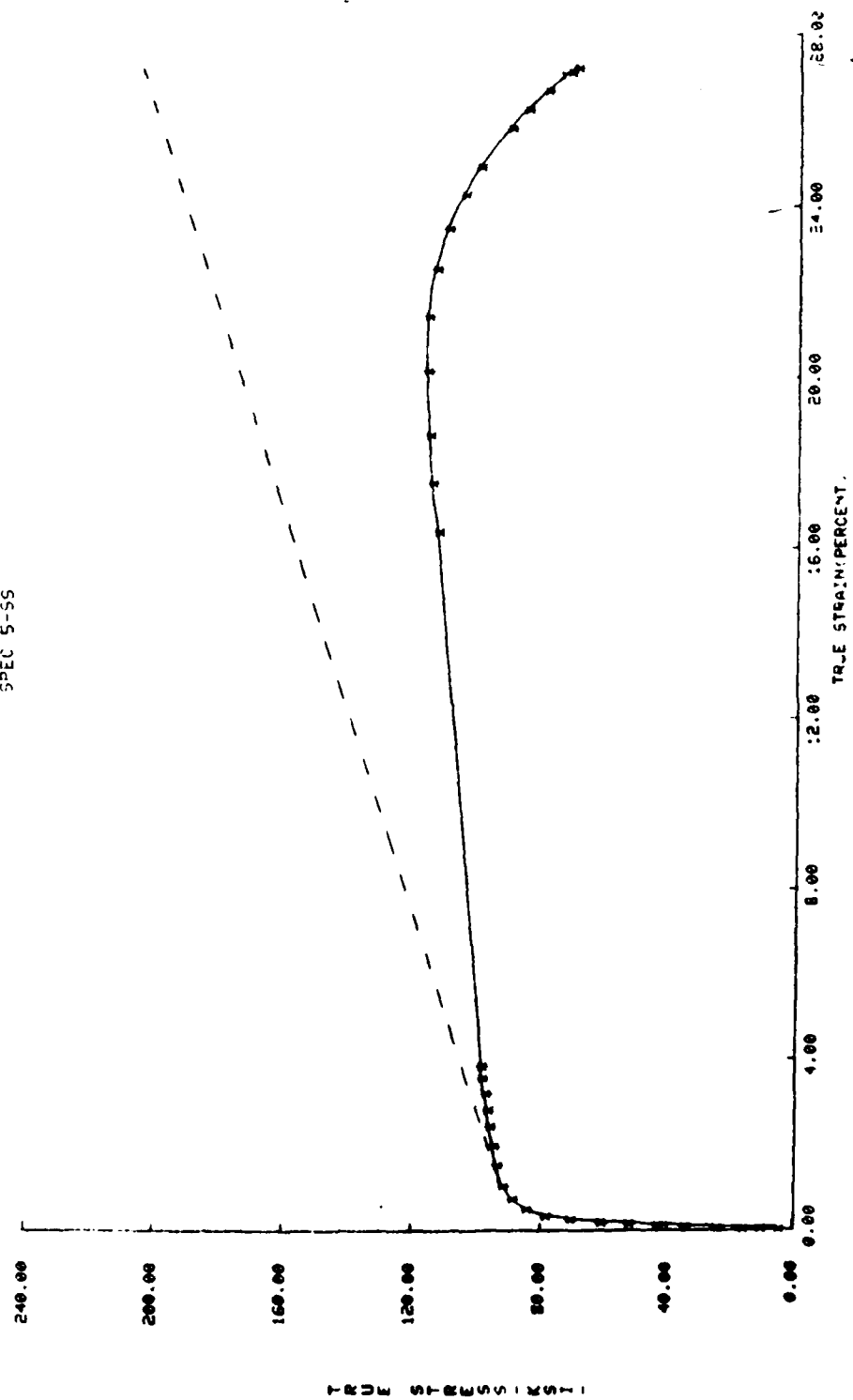
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SPEC 3-TI



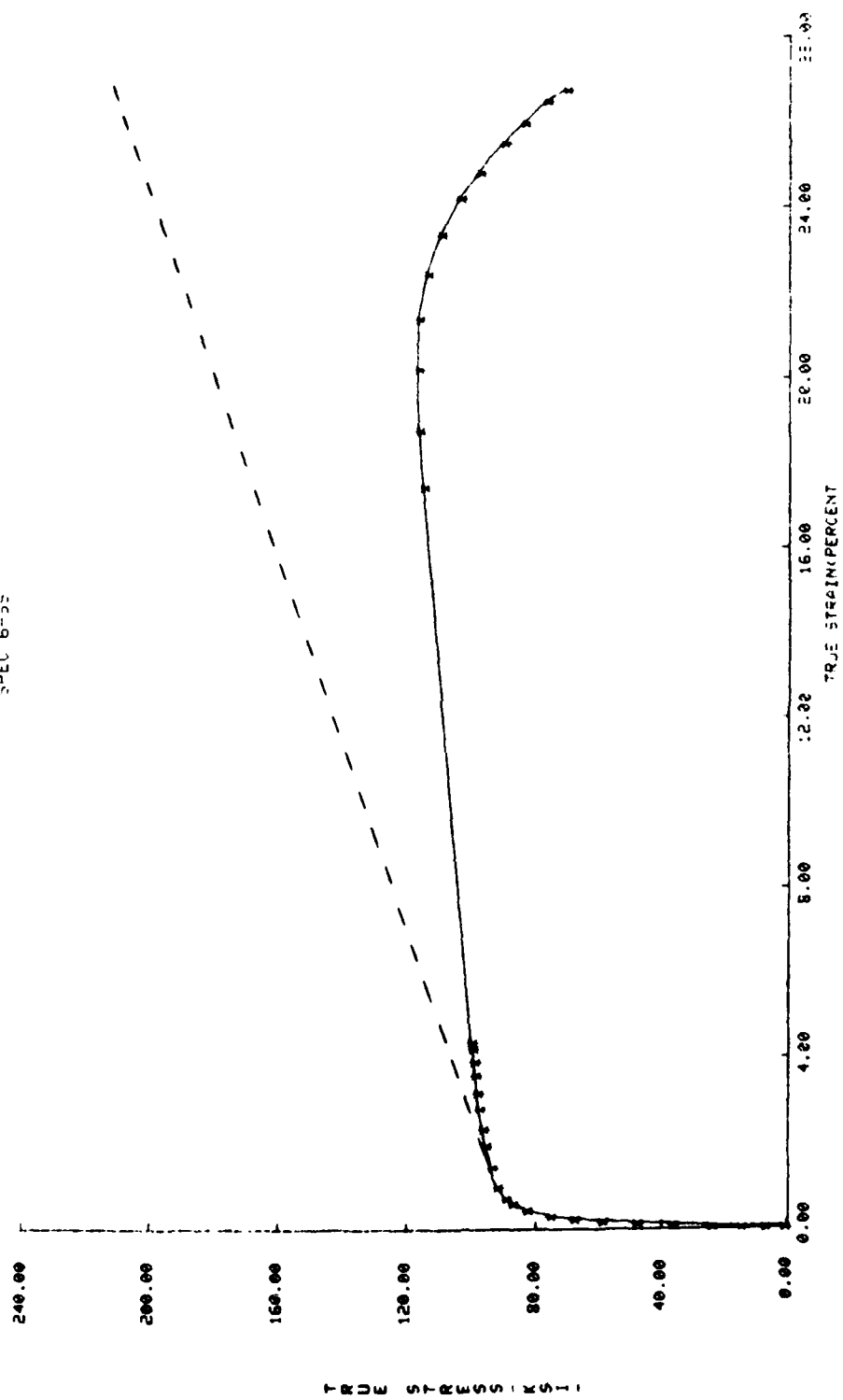
STANDARD TENSILE  
3.001 STRAIN SEC  
412 THINNESS STEEL  
SPEC 4-95



STANDARD TENSILE  
0.001 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 5-55



STANDARD TENSILE  
0.001 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 6-39



STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-IMO-1U TITANIUM  
SPEC 4-TI

1	.0335636	3.257
2	.0900508	10.586
3	.15638171	19.547
4	.21490247	27.294
5	.25340253	35.854
6	.3139742	45.468
7	.3825160	57.491
8	.42774387	65.659
9	.46604360	75.053
10	.54401987	84.065
11	.60184056	95.112
12	.6460636	100.849
13	.65938212	107.800
14	.7177952	114.372
15	.7365080	121.735
16	.75545377	127.467
17	.81327086	137.725
18	.87152658	144.720
19	.89032801	151.276
20	.94877471	156.643
21	1.12597012	158.445
22	1.36217390	157.985
23	1.59776250	157.962
24	1.93054144	158.097
25	2.4268198	158.611
26	2.57369316	160.402
27	3.13423510	163.428
28	3.71172878	164.412
29	4.26676334	165.786
30	4.79374945	167.132
31	5.34901999	166.809
32	5.97021255	167.891
33	6.43831310	168.280
34	7.07130284	169.827
35	7.73724689	171.008
36	8.38344017	171.278
37	9.47307752	171.030
38	10.19656066	171.017
39	10.73485088	170.518
40	11.28830543	169.227
41	11.89194646	167.491
42	12.36882362	164.450
43	12.32591671	161.153
44	13.24538632	157.787
45	13.24538632	154.789

STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-IMO-1U TITANIUM  
SPEC 5-TI

1	.05654797	3.271
2	.10870553	11.008

3	.1265188	17.106
4	.19257119	36.831
5	.24475782	34.641
6	.29444439	44.823
7	.34450159	54.625
8	.37627846	62.781
9	.42550477	73.394
10	.47620597	82.388
11	.52625225	92.199
12	.55599913	102.494
13	.58810322	110.171
14	.62098541	117.127
15	.65267456	125.307
16	.70451609	132.898
17	.73736615	140.070
18	.77177377	145.410
19	.82670694	149.961
20	.82355972	153.219
21	.94080533	154.591
22	1.07879180	154.766
23	1.21697999	154.611
24	1.35497748	154.473
25	1.5704257	155.306
26	1.98085383	156.529
27	2.31140082	158.013
28	2.67925197	159.996
29	3.10483700	160.861
30	3.4703574	161.809
31	3.91194067	162.936
32	4.44666631	164.045
33	4.96035564	164.209
34	5.52645895	166.310
35	6.22230569	166.499
36	6.65211026	166.982
37	7.33945767	168.020
38	7.91215402	168.381
39	8.37140514	168.936
40	8.73736618	169.290
41	9.17541029	168.917
42	9.70158121	169.176
43	10.17083419	169.756
44	10.56659295	169.280
45	11.03284015	168.501
46	11.56673857	169.221
47	12.15177187	168.690
48	12.69868228	167.807
49	13.22621626	165.083
50	13.62908264	162.723
51	14.08292549	159.521
52	14.44755739	157.500
53	15.24390454	149.787
54	15.55330532	147.591
55	15.81033547	145.748
56	15.96525145	143.217

STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-IMO-1U TITANIUM  
SPEC 6-TI



1	0.3471539	3.667	59	15.31320832	142.215
2	.06552873	9.359		STANDARD TENSILE	
3	.13416839	17.916		0.100 STRAIN/SEC	
4	.17952807	28.084		4.00 STAINLESS STEEL	
5	.22423013	38.663		SPEC 9-55	
6	.27366970	46.410	1	.00035853	.203
7	.32124898	54.557	2	.00896298	5.063
8	.36710015	64.350	3	.03366054	13.368
9	.47082861	75.415	4	.0358711	20.356
10	.51661125	85.211	5	.08648913	28.339
11	.56697375	91.767	6	.09795168	32.825
12	.59804342	97.897	7	.15035226	39.928
13	.6282871	104.436	8	.12266335	46.814
14	.69349202	114.679	9	.13340555	52.898
15	.74032105	123.685	10	.18185101	57.777
16	.80693294	132.722	11	.14915893	61.821
17	.87464714	140.954	12	.19723884	66.500
18	.88640938	146.669	13	.24708391	72.197
19	.91967226	151.195	14	.29690415	77.897
20	.93372116	155.287	15	.38147019	80.774
21	1.12844386	157.609	16	.38789905	84.431
22	1.22766277	157.348	17	.47274506	87.516
23	1.32621259	157.500	18	.91840081	93.369
24	1.46515929	156.985	19	.127692285	95.530
25	1.64319673	156.553	20	1.00832474	97.753
26	1.93733236	157.193	21	2.16864879	100.011
27	2.17145361	158.115	22	3.58243795	101.836
28	2.42451498	159.084	23	4.46534833	103.329
29	2.67637448	160.467	24	5.40178148	104.984
30	3.14007390	162.747	25	6.54450526	106.257
31	3.44913410	163.444	26	7.28915843	106.794
32	3.98770165	164.666	27	8.21265215	107.736
33	4.42849011	165.253	28	9.23323846	108.827
34	4.7222466	165.620	29	10.3420005	109.543
35	5.22852417	166.249	30	11.46674946	109.983
36	5.68275105	166.876	31	12.42284052	110.755
37	6.2099722	167.677	32	13.12544181	111.495
38	6.62237795	167.785	33	13.95556107	111.487
39	6.95872451	168.574	34	14.93570225	111.608
40	7.42317698	169.673	35	15.67929211	111.385
41	7.94188394	170.038	36	16.52665221	110.611
42	8.40233466	170.702	37	17.26618215	109.703
43	8.76981055	170.706	38	17.86552894	108.392
44	9.22700084	170.925	39	18.66132649	106.526
45	9.69907468	171.406	40	19.4867614	103.633
46	9.95512767	170.467	41	20.13318117	101.580
47	10.40634295	170.669	42	20.81303257	99.488
48	10.88246422	171.183	43	21.32688160	96.964
49	11.19746146	171.797	44	21.97225420	93.540
50	11.66147829	171.028	45	22.72621590	89.606
51	12.05515430	168.814	46	23.48061967	84.863
52	12.53385115	168.157	47	24.04264046	81.188
53	13.10192704	164.035	48	24.50812256	77.659
54	13.36780151	161.452	49	24.97889934	73.575
55	13.80852118	157.880	50	25.21722951	70.356
56	14.24730702	154.271	51	25.49174514	67.922
57	14.59795432	150.426	52	25.76644683	64.959
58	14.98125574	147.095		STANDARD TENSILE	

0.100 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 10-SS

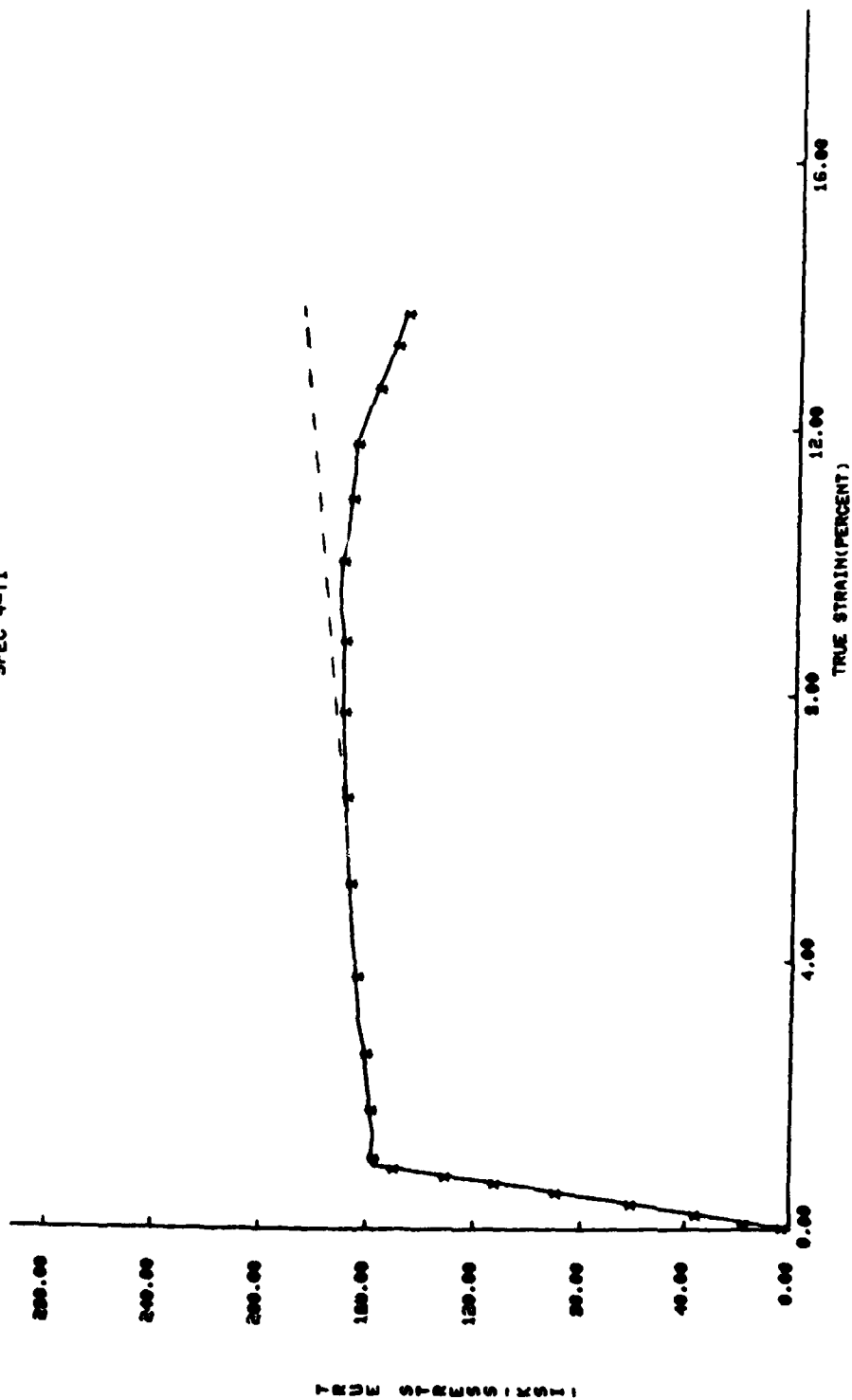
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2	.01210789	3.239
3	.04877515	8.310
4	.06218912	15.403
5	.10791047	24.531
6	.11829335	32.438
7	.17355414	39.144
8	.18116430	47.661
9	.19682618	54.156
10	.17567809	59.823
11	.15754759	64.680
12	.22436764	68.160
13	.24757130	72.633
14	.27001492	77.311
15	.37350818	81.631
16	.47840062	85.550
17	.58468773	89.083
18	.61268690	91.957
19	1.12061854	94.519
20	1.54953608	96.403
21	2.20984863	98.348
22	3.96558705	100.354
23	3.63792577	101.586
24	4.25027029	102.695
25	5.06670736	104.071
26	5.95452543	105.286
27	6.76298396	106.375
28	7.59534481	107.210
29	8.58904225	108.393
30	9.39285159	108.917
31	10.19027116	109.435
32	11.01644387	110.216
33	11.86240655	110.271
34	12.85692657	111.073
35	14.05325846	110.738
36	15.09070747	111.087
37	16.15403901	110.515
38	17.31074385	108.593
39	18.15262109	107.204
40	18.85661755	105.132
41	19.82161320	101.853
42	20.58144965	98.768
43	21.17300813	95.044
44	22.4384566	92.864
45	23.12257405	89.268
46	24.11791677	85.141
47	24.50115880	81.819
48	24.81950740	75.177
49	25.13743427	72.618
50	25.32935102	69.782
51	25.48800529	67.343
52		65.390
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STANDARD TENSILE  
0.100 STRAIN/SEC

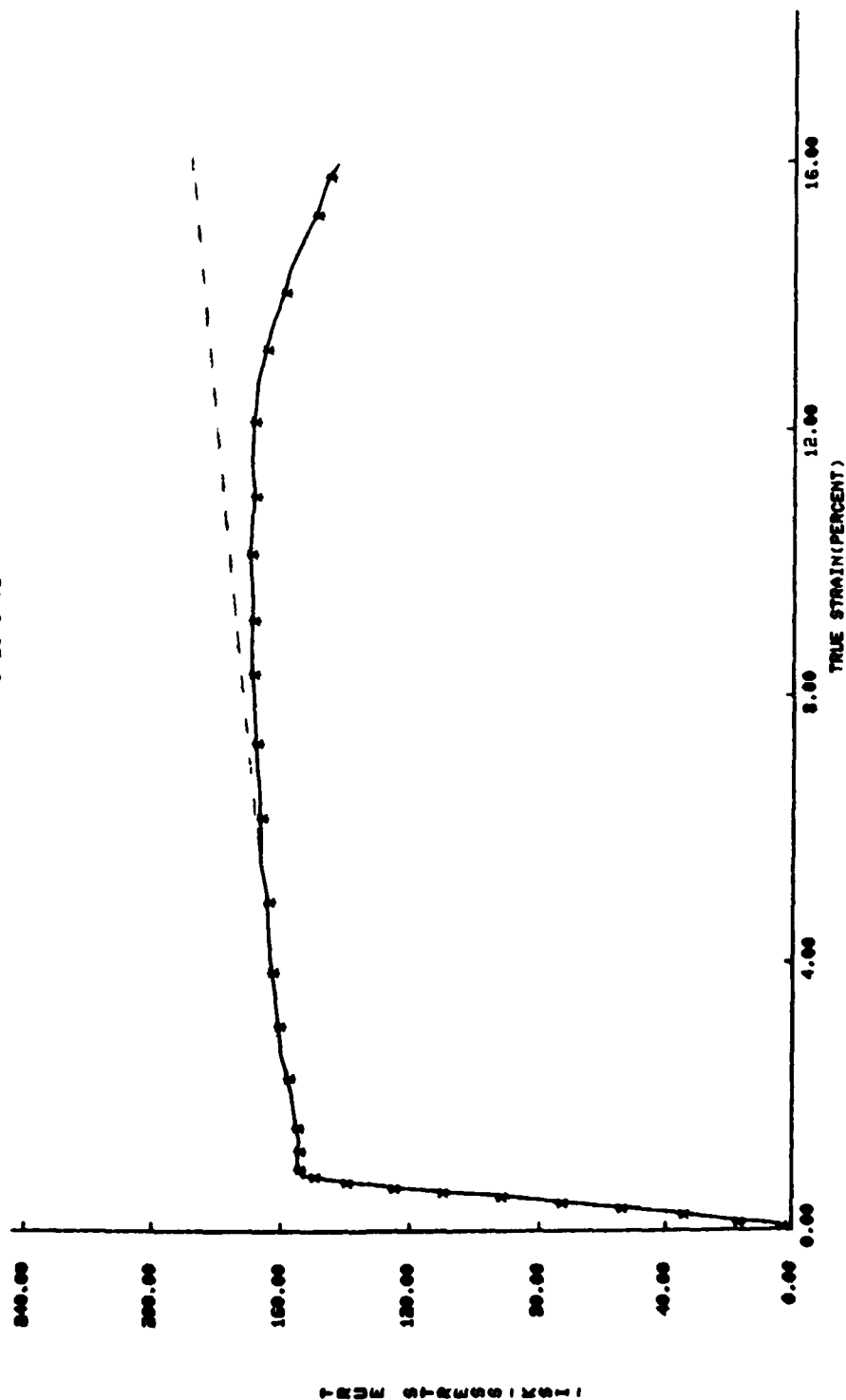
410 STAINLESS STEEL  
SPEC 11-SS

1	0.00000000	.608
2	.03991219	7.090
3	.03991219	12.155
4	.05808846	18.237
5	.07980846	24.113
6	.07980846	30.800
7	.11968882	37.293
8	.11968882	42.968
9	.15955328	48.140
10	.15955328	53.722
11	.15955328	58.588
12	.19940186	61.831
13	.23923456	66.510
14	.27005141	71.596
15	.27005141	74.859
16	.27905141	77.659
17	.31885240	80.761
18	.31885240	83.350
19	.35863756	86.037
20	.47789816	88.290
21	.71593351	90.383
22	1.11153265	92.814
23	1.78046246	95.634
24	2.40595317	98.539
25	3.22101687	99.779
26	4.33575991	101.639
27	5.2490826	103.622
28	6.45385211	105.212
29	8.08714752	106.487
30	9.51287081	108.021
31	8.7111724	103.139
32	10.30824738	109.350
33	11.24015376	110.000
34	11.80934776	110.351
35	12.69227774	110.526
36	13.91542771	110.589
37	14.88328870	110.091
38	15.94403432	108.582
39	16.62246298	107.667
40	17.56459646	106.431
41	18.06568319	103.372
42	18.53110956	101.925
43	19.22521250	99.239
44	20.14335173	96.011
45	21.79337212	92.644
46	21.5606980	88.322
47	21.58647288	84.247
48	22.56155433	80.822
49	23.0066460	76.994
50	23.3232038	73.003
51	23.5448534	70.427
52	23.73238038	68.282
53		66.105

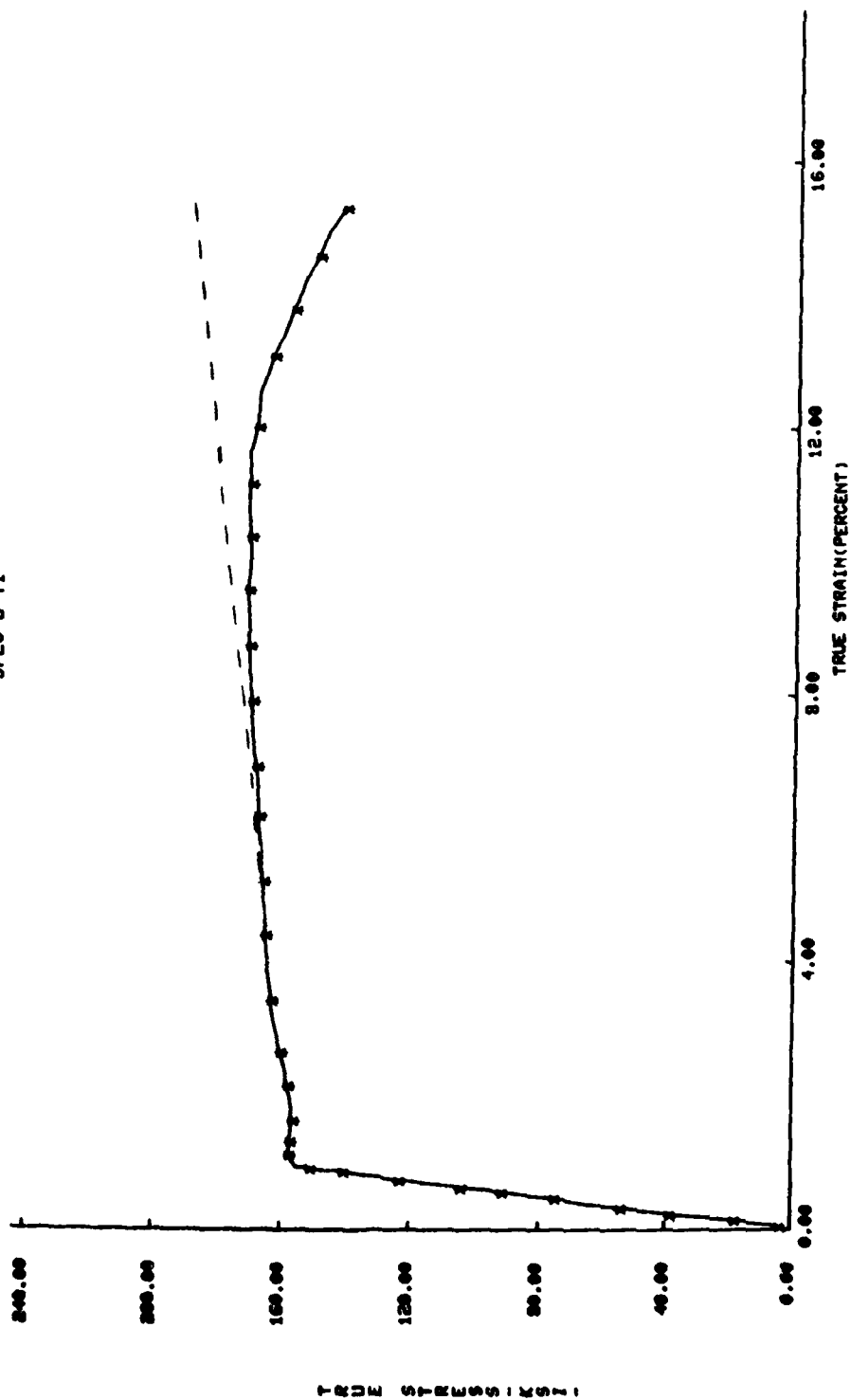
STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-INO-LU TITANIUM  
SPEC 4-TI



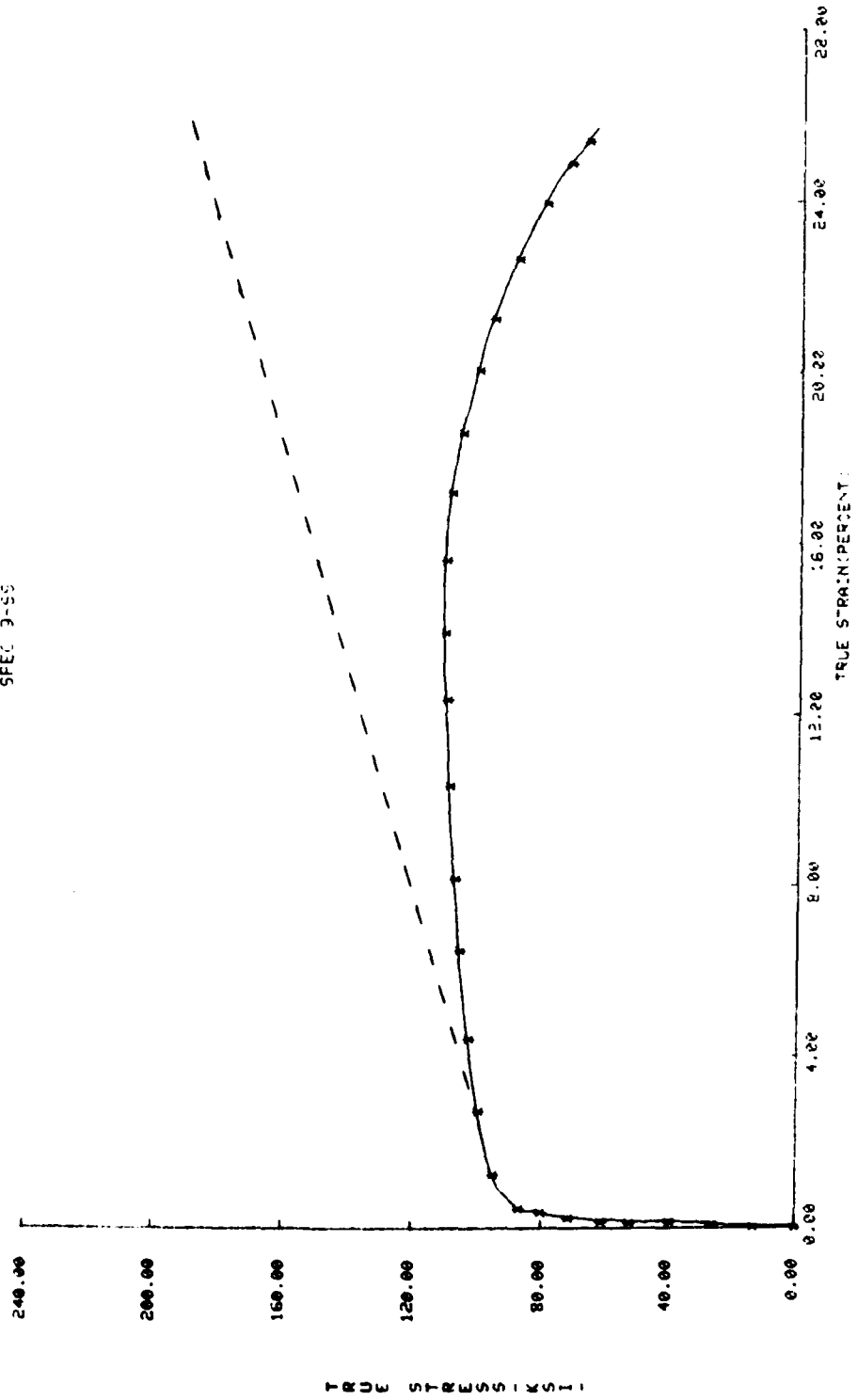
STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-100-10 TITANIUM  
SPEC 5-TI



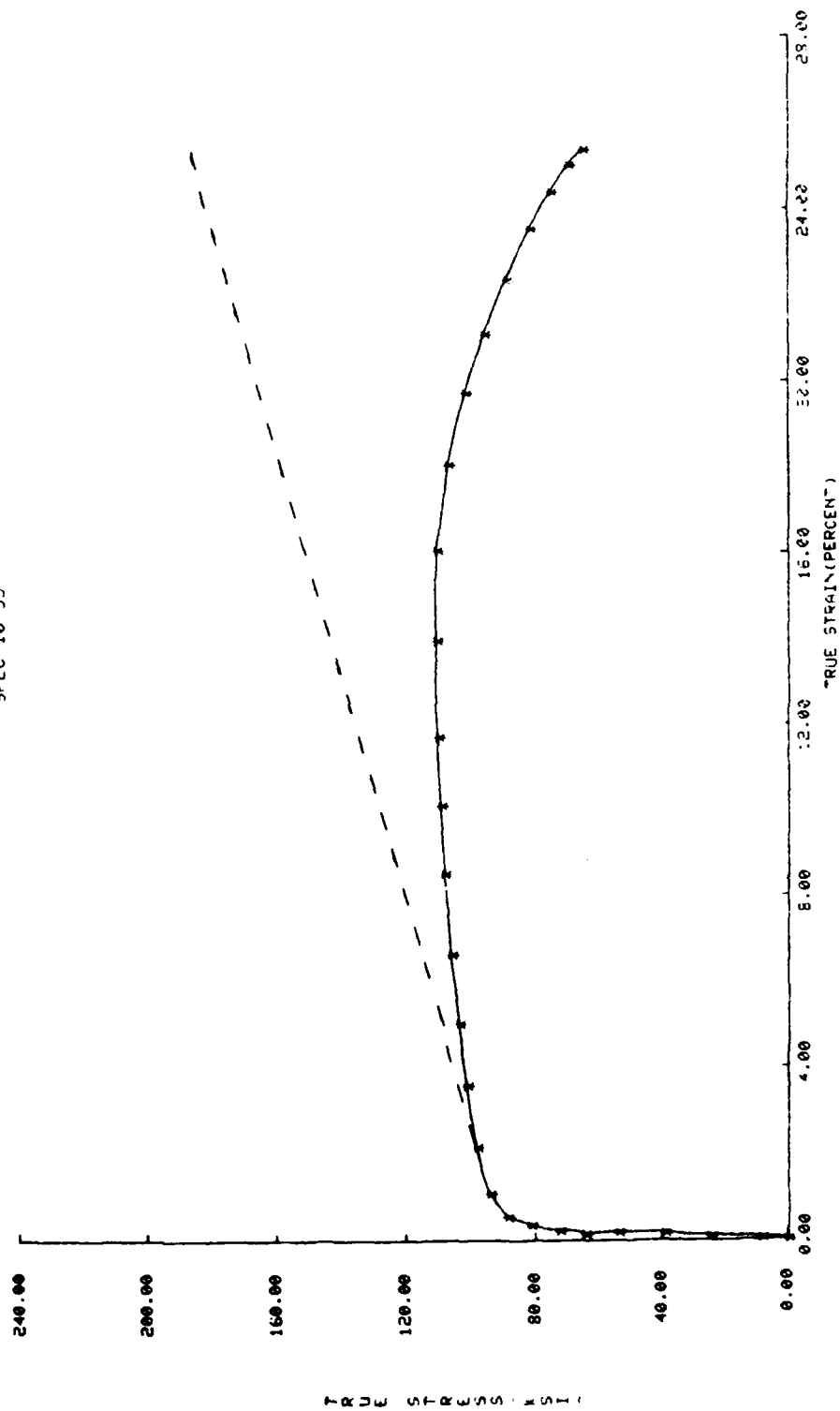
STANDARD TENSILE  
0.100 STRAIN/SEC  
BAL-100-1U TITANIUM  
SPEC 6-TI



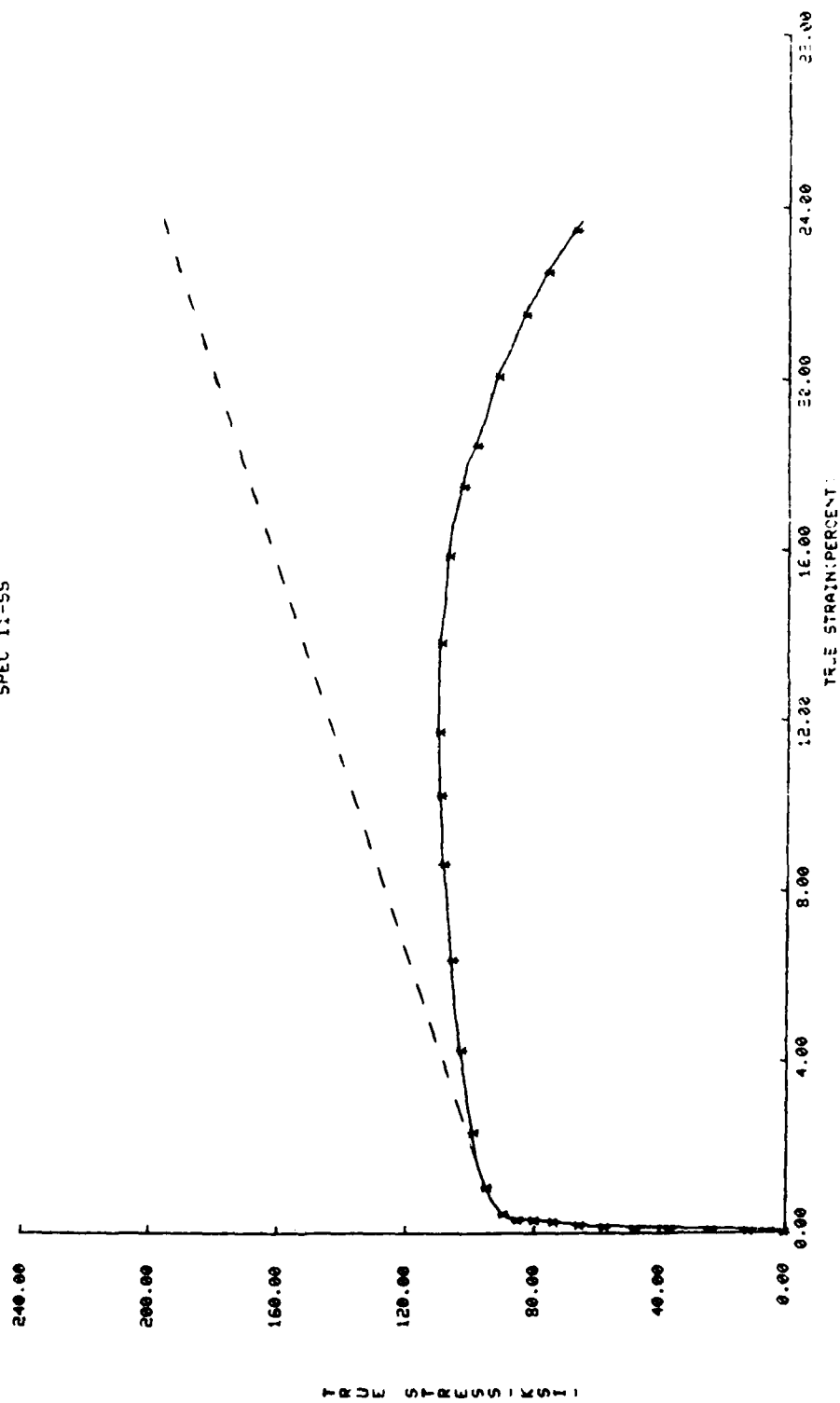
STANDARD TENSILE  
0.100 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 3-55



STANDARD TENSILE  
0.100 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 10-55



STANDARD TENSILE  
0.100 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 11-55





STANDARD TENSILE  
1 STRAIN/SEC  
410 STAINLESS STEEL  
12-55

0	0.0000000
1	.03375393
2	.02970882
3	.01959532
4	.0533264
5	.04120919
6	.07292752
7	.06551426
8	.05742644
9	.08946582
10	.07768440
11	.07364106
12	.06555391
13	.10063317
14	.09456950
15	.08783165
16	.12155595
17	.11414630
18	.10538873
19	.13843365
20	.12900468
21	.12092200
22	.14857474
23	.17891266
24	.16746771
25	.15938814
26	.19106900
27	.21937528
28	.20658896
29	.19851255
30	.23018102
31	.21806902
32	.24838586
33	.23829446
34	.23291196
35	.27062292
36	.38568597
37	.45963599
38	.61273456
39	.76559909
40	.99779722
41	1.30811156
42	1.73562019
43	2.12366414
44	2.54903252
45	3.08982778
46	3.62771417
47	4.23781135
48	5.03439341
49	6.03918727
50	6.24033942
51	

52	7.17148336
53	7.76414532
54	8.27815152
55	8.97236890
56	9.29953415
57	9.84244214
58	10.38302644
59	10.92070014
60	11.45618771
61	12.02458109
62	12.58865185
63	13.08087454
64	13.64021175
65	14.09947266
66	14.85378838
67	15.40448956
68	16.12409898
69	16.73751559
70	17.38186599
71	17.95382701
72	18.42432343
73	18.76063940
74	19.32868171
75	19.82765279
76	20.22273092
77	20.58658465
78	20.91571847
79	21.40739690
80	21.73438726
81	22.25395437
82	22.6495784
83	23.06446173
84	23.35616003
85	23.54944340
86	23.80462585
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STANDARD TENSILE  
1 STRAIN/SEC  
410 STAINLESS STEEL  
13-55

1	.00166925
2	.01085061
3	.01983012
4	.0898166
5	.05220655
6	.01545909
7	.04529750
8	.03111444
9	.05510926
10	.07159243
11	.08557121
12	.10621810
13	.09004367
14	.11185555
15	.1283100
16	.1563420
17	.1393355
18	.1593573
19	.18190014
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107.92:
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68.365
66.774

20	16190374	77.177	39772422	90.963
21	19536350	78.621	42745227	93.876
22	18836637	80.450	50752756	93.684
23	17836932	82.887	62076572	95.407
24	26649340	85.755	77486389	96.756
25	27270646	89.055	1.04573568	98.724
26	45549639	93.046	1.56238006	100.716
27	64707478	94.834	2.25916111	103.355
28	1.83326439	97.498	2.71474703	104.506
29	1.37948193	99.931	3.4854754	105.558
30	1.88366488	101.939	4.4805992	107.392
31	2.81238251	104.455	4.8037580	107.604
32	3.73341684	106.791	5.12429817	109.005
33	4.7237480	108.850	6.10425811	109.179
34	5.44437738	109.478	7.0190261	111.118
35	6.53241432	111.024	7.82246464	111.853
36	7.38916581	111.648	8.29434075	112.411
37	8.27303800	112.970	8.91620445	113.180
38	9.78625555	113.395	9.53422489	113.950
39	9.58853940	113.741	10.32903312	114.723
40	10.20386748	114.074	10.94097506	114.591
41	11.06489767	114.947	11.58152174	114.628
42	12.06664200	115.592	12.11462387	115.621
43	12.94370767	115.768	12.43198903	116.026
44	14.20827862	116.239	12.96058844	116.010
45	15.20113769	116.697	13.76472168	116.345
46	16.12637827	116.157	14.18260728	116.181
47	17.07674931	115.724	15.11420691	116.675
48	18.25590036	113.866	15.87031480	115.752
49	19.25859820	113.866	16.6906251	114.172
50	20.18654113	110.743	17.60530250	111.952
51	20.98059762	106.959	18.21123596	110.277
52	21.83011638	103.431	18.57906909	109.511
53		99.657	19.31541953	105.988
54			19.87960730	103.694
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56			20.99988239	98.508
57			21.45859233	95.756
58			21.85249242	92.402
59			22.24312869	89.778
60			22.72814785	86.973
61			23.05265404	83.981
62			23.53546838	80.350
63			23.85962393	76.279
64			24.21178440	73.240
65			24.50087974	69.866
66			24.63177085	67.121
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08314563	0740583	10511511	13531527	16558013	15788924	19032058	22299920	21441315	24636861	27724123	30823255	33855526	36924444	40238774	43190365	46115118	49279759	52177041	552144962	583530482	614661040	646944344	678876979	71048149	742340428	774594594	806514684	83823764	87018223	902358224	934839318	9674020904	1000240222	103224109	1064093579	1095931366	11274751395	11591357156	119082395	1222407072	12531767822	1283718251	131426724	134463821	137462617	140425592	143394970	1463737579	149366673	1523729021	1553936457	1584290583	1614813802	16455012403			
20.726	25.388	30.890	36.384	41.270	46.146	50.425	54.583	59.380	63.866	68.964	73.232	77.519	82.012	84.248	84.923	88.404	90.561	92.343	94.169	96.246	97.837	99.040	101.045	102.622	104.126	105.473	107.254	108.872	108.557	109.756	110.656	111.371	112.338	112.864	113.087	113.432	113.478	113.812	114.249	114.423	114.228	114.505	114.048	113.175	110.818	109.245	106.927	102.087	102.685	100.141	75.180	72.027	69.663	67.428	65.381		
0.1983832	0.1904327	0.57611565	0.7625065	1.160240	1.5244320	2.207361	2.6667286	3.2363972	3.8037609	4.5222789	5.1378965	5.9014377	6.4702546	7.0407222	7.659323	8.1777244	8.3616776	8.5485537	8.7363802	8.9310676	9.17173135	1.22820881	1.54285478	2.01287627	2.51959359	3.08177673	3.54473286	4.08205791	4.42592879	4.76871603	5.11023780	5.41130865	5.77145139	6.12851461	6.40962166	6.74559444	7.19174777	7.61753604	8.06028316	8.5001347	8.92115808	9.35809238	9.79294664	10.29816654	10.83636932	11.24721632	11.60330874	12.06361657	12.38123761	12.71541340	13.06623678	13.38104198	13.83438360	14.14712880			
1 STRAIN/SEC	BAL-IMO-1U-TITANIUM	7-11	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55
STANDARD TENSILE																																																									

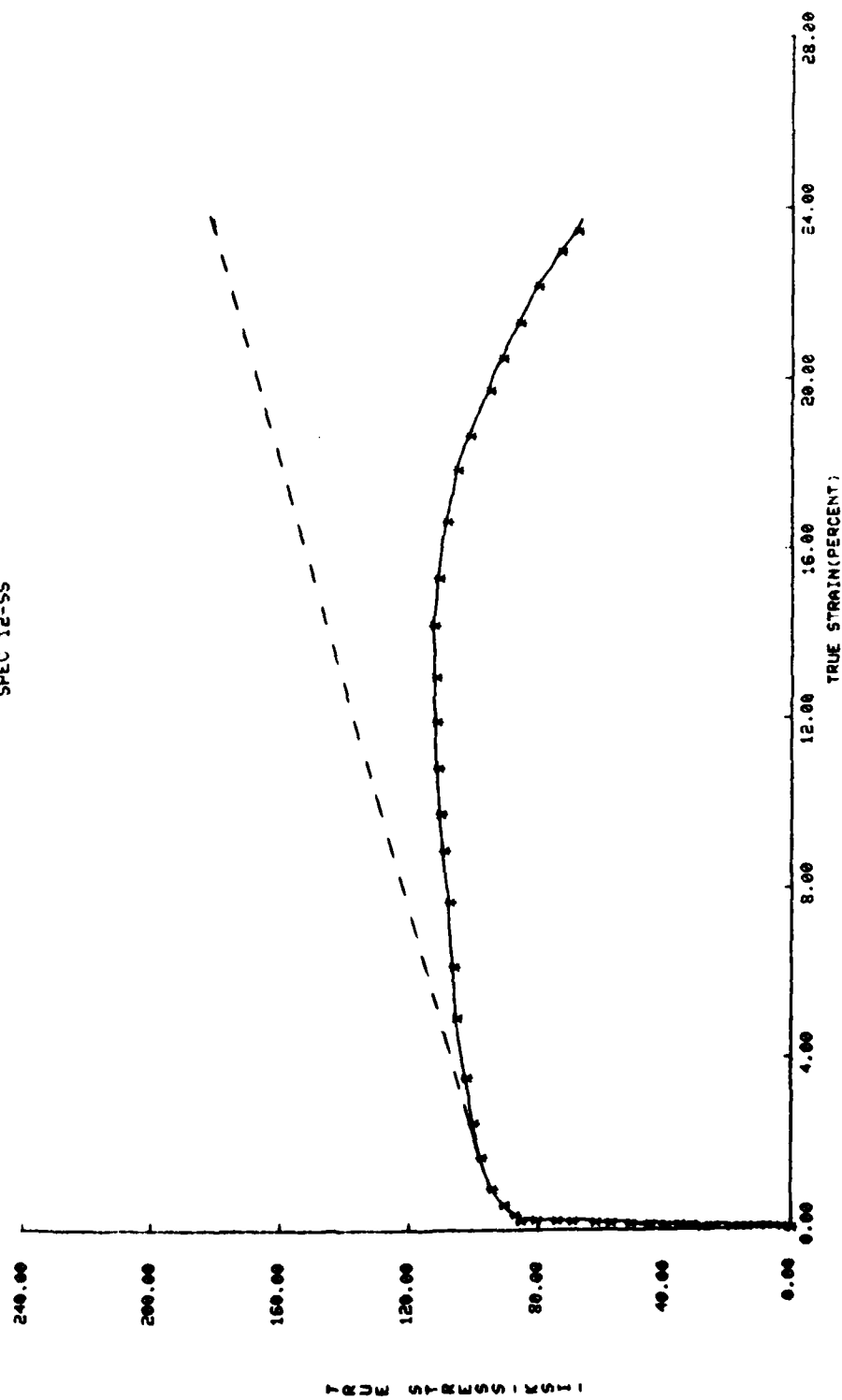
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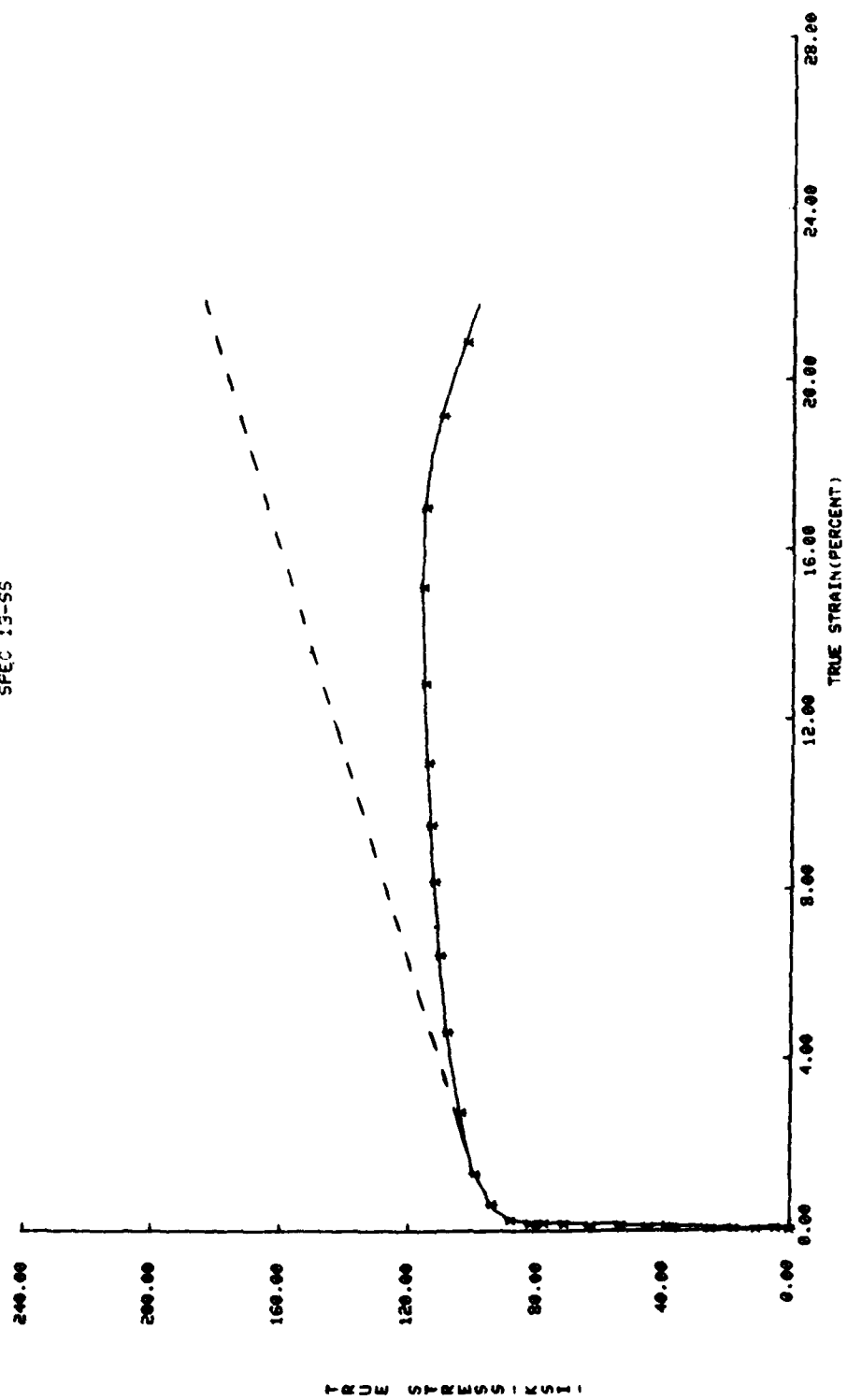
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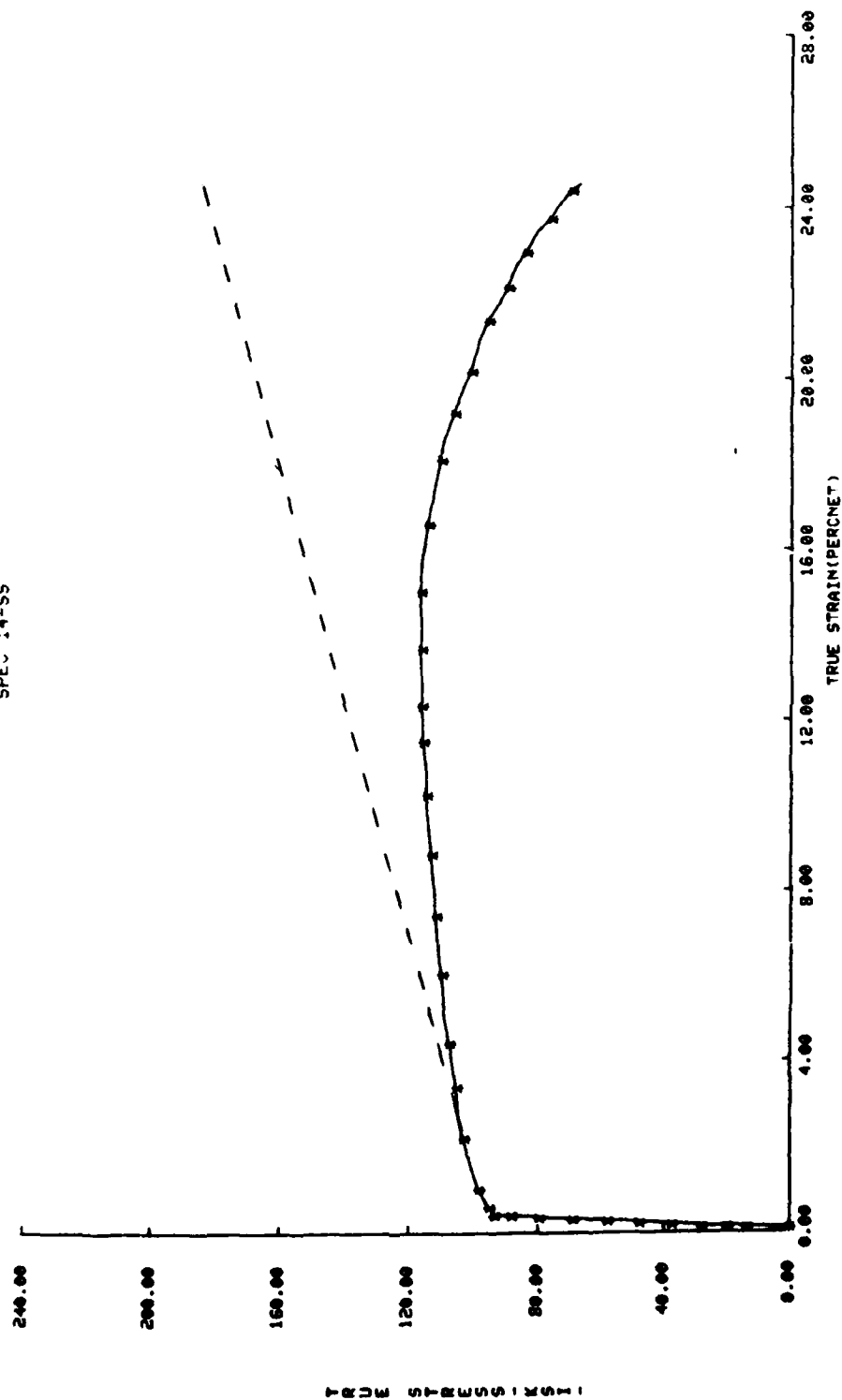
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410 STAINLESS STEEL  
SPEC 12-55



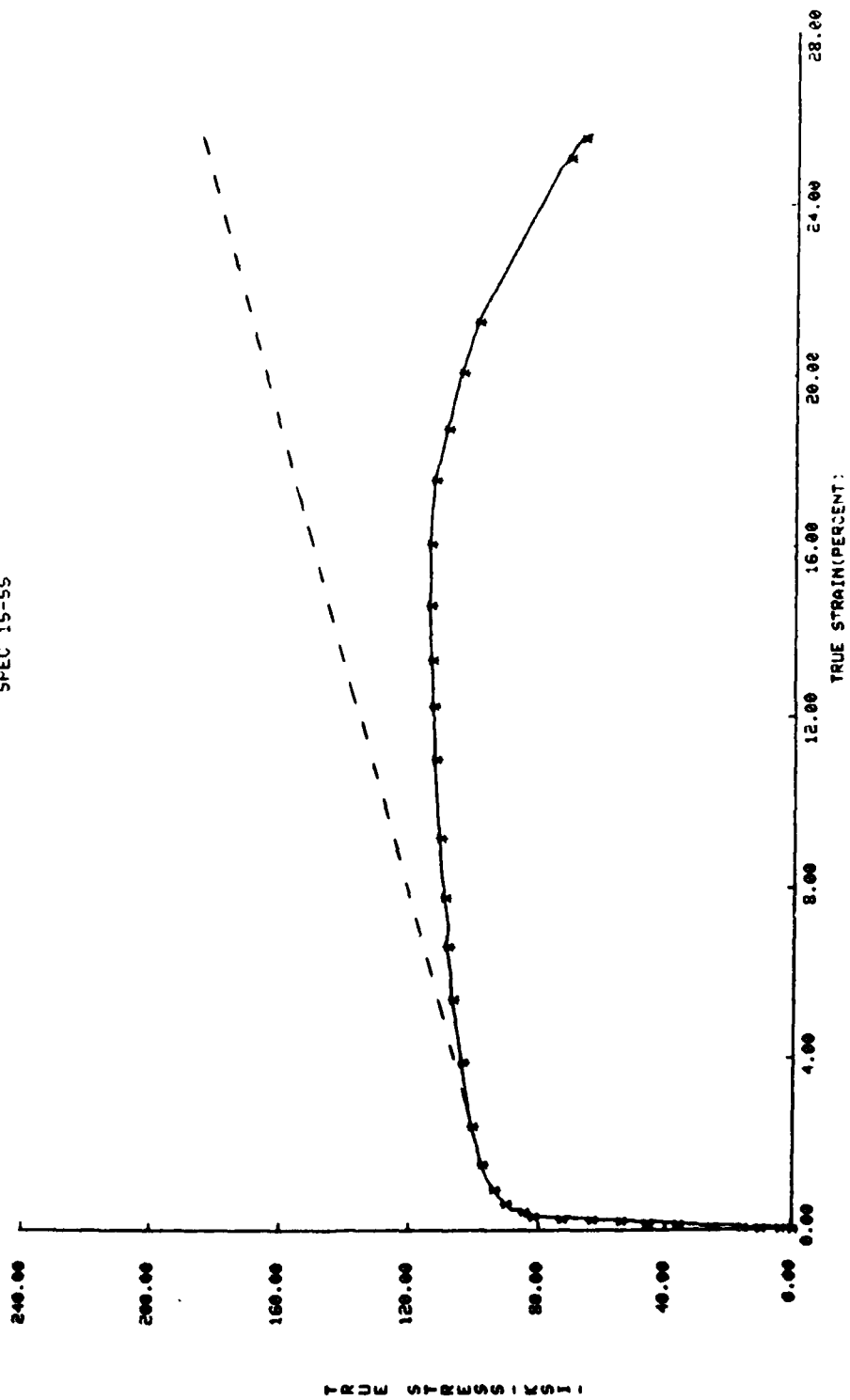
STANDARD TENSILE  
1 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 13-SS



STANDARD TENSILE  
1 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 14-SS

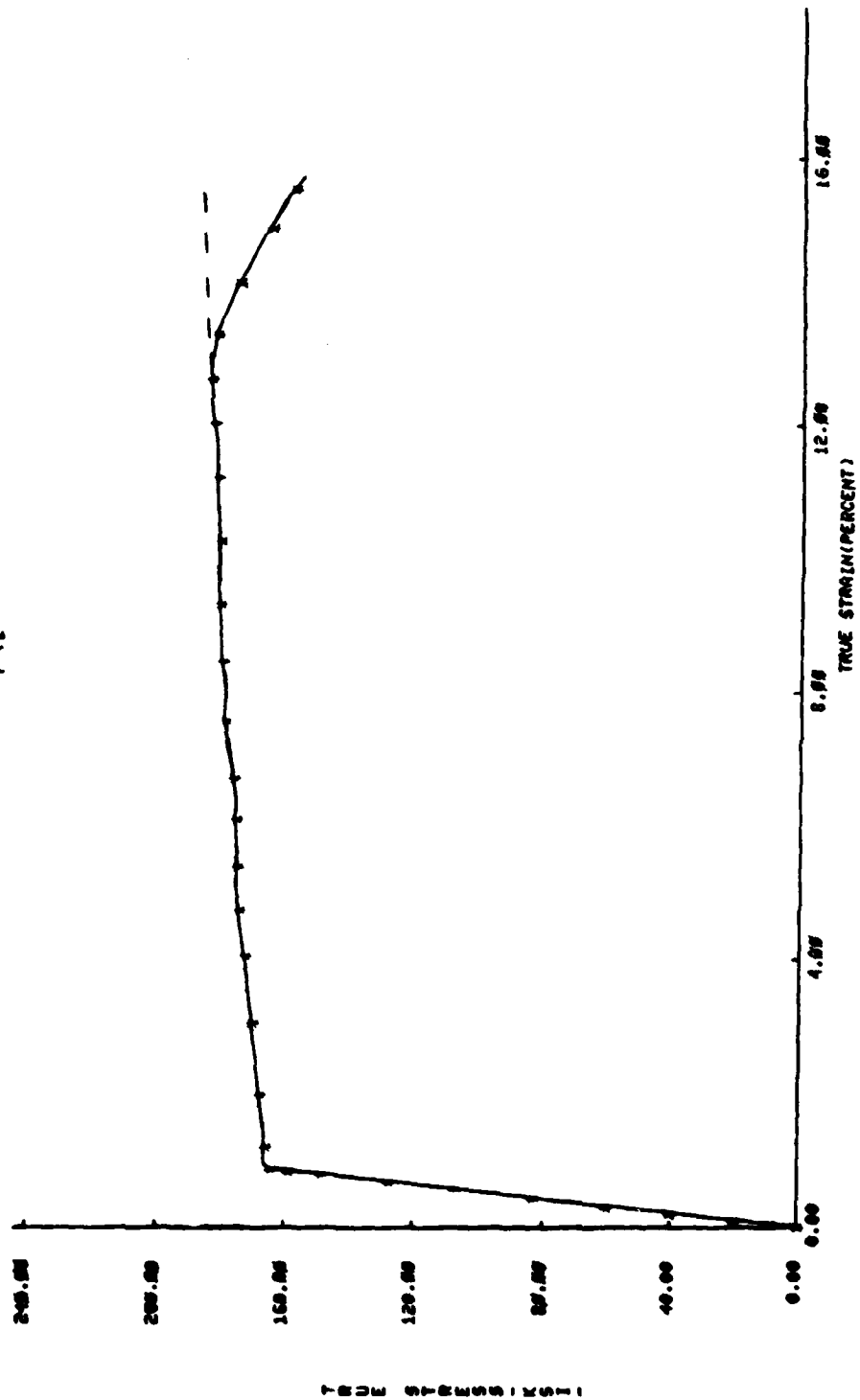


STANDARD TENSILE  
1 STRAIN/SEC  
410 STAINLESS STEEL  
SPEC 15-55





STANDARD TENSILE  
 1 STRAIN/SEC  
 8AL-100-10-TITANIUM  
 7-11



APPENDIX B  
HIGH STRAIN RATE CURVES AND  
DIGITAL DATA



1	1.0407057	191.853
2	1.0407116	192.814
3	1.0407175	193.775
4	1.0407234	194.736
5	1.0407293	195.697
6	1.0407352	196.658
7	1.0407411	197.619
8	1.0407470	198.580
9	1.0407529	199.541
10	1.0407588	200.502
11	1.0407647	201.463
12	1.0407706	202.424
13	1.0407765	203.385
14	1.0407824	204.346
15	1.0407883	205.307
16	1.0407942	206.268
17	1.0408001	207.229
18	1.0408060	208.190
19	1.0408119	209.151
20	1.0408178	210.112
21	1.0408237	211.073
22	1.0408296	212.034
23	1.0408355	212.995
24	1.0408414	213.956
25	1.0408473	214.917
26	1.0408532	215.878
27	1.0408591	216.839
28	1.0408650	217.800
29	1.0408709	218.761
30	1.0408768	219.722
31	1.0408827	220.683
32	1.0408886	221.644
33	1.0408945	222.605
34	1.0409004	223.566
35	1.0409063	224.527
36	1.0409122	225.488
37	1.0409181	226.449
38	1.0409240	227.410
39	1.0409299	228.371
40	1.0409358	229.332
41	1.0409417	230.293
42	1.0409476	231.254
43	1.0409535	232.215
44	1.0409594	233.176
45	1.0409653	234.137
46	1.0409712	235.098
47	1.0409771	236.059
48	1.0409830	237.020
49	1.0409889	237.981
50	1.0409948	238.942
51	1.0410007	239.903
52	1.0410066	240.864
53	1.0410125	241.825
54	1.0410184	242.786
55	1.0410243	243.747
56	1.0410302	244.708
57	1.0410361	245.669
58	1.0410420	246.630
59	1.0410479	247.591
60	1.0410538	248.552
61	1.0410597	249.513
62	1.0410656	250.474
63	1.0410715	251.435
64	1.0410774	252.396
65	1.0410833	253.357
66	1.0410892	254.318
67	1.0410951	255.279
68	1.0411010	256.240
69	1.0411069	257.201
70	1.0411128	258.162
71	1.0411187	259.123
72	1.0411246	260.084
73	1.0411305	261.045
74	1.0411364	262.006
75	1.0411423	262.967
76	1.0411482	263.928
77	1.0411541	264.889
78	1.0411600	265.850
79	1.0411659	266.811
80	1.0411718	267.772
81	1.0411777	268.733
82	1.0411836	269.694
83	1.0411895	270.655
84	1.0411954	271.616
85	1.0412013	272.577
86	1.0412072	273.538
87	1.0412131	274.499
88	1.0412190	275.460
89	1.0412249	276.421
90	1.0412308	277.382
91	1.0412367	278.343
92	1.0412426	279.304
93	1.0412485	280.265
94	1.0412544	281.226
95	1.0412603	282.187
96	1.0412662	283.148
97	1.0412721	284.109
98	1.0412780	285.070
99	1.0412839	286.031
100	1.0412898	286.992
101	1.0412957	287.953
102	1.0413016	288.914
103	1.0413075	289.875
104	1.0413134	290.836
105	1.0413193	291.797
106	1.0413252	292.758
107	1.0413311	293.719
108	1.0413370	294.680
109	1.0413429	295.641
110	1.0413488	296.602
111	1.0413547	297.563
112	1.0413606	298.524
113	1.0413665	299.485
114	1.0413724	300.446
115	1.0413783	301.407
116	1.0413842	302.368
117	1.0413901	303.329
118	1.0413960	304.290
119	1.0414019	305.251
120	1.0414078	306.212
121	1.0414137	307.173
122	1.0414196	308.134
123	1.0414255	309.095
124	1.0414314	310.056
125	1.0414373	311.017
126	1.0414432	311.978
127	1.0414491	312.939
128	1.0414550	313.900
129	1.0414609	314.861
130	1.0414668	315.822
131	1.0414727	316.783
132	1.0414786	317.744
133	1.0414845	318.705
134	1.0414904	319.666
135	1.0414963	320.627
136	1.0415022	321.588
137	1.0415081	322.549
138	1.0415140	323.510
139	1.0415199	324.471
140	1.0415258	325.432
141	1.0415317	326.393
142	1.0415376	327.354
143	1.0415435	328.315
144	1.0415494	329.276
145	1.0415553	330.237
146	1.0415612	331.198
147	1.0415671	332.159
148	1.0415730	333.120
149	1.0415789	334.081
150	1.0415848	335.042
151	1.0415907	336.003
152	1.0415966	336.964
153	1.0416025	337.925
154	1.0416084	338.886
155	1.0416143	339.847
156	1.0416202	340.808
157	1.0416261	341.769
158	1.0416320	342.730
159	1.0416379	343.691
160	1.0416438	344.652
161	1.0416497	345.613
162	1.0416556	346.574
163	1.0416615	347.535
164	1.0416674	348.496
165	1.0416733	349.457
166	1.0416792	350.418
167	1.0416851	351.379
168	1.0416910	352.340
169	1.0416969	353.301
170	1.0417028	354.262
171	1.0417087	355.223
172	1.0417146	356.184
173	1.0417205	357.145
174	1.0417264	358.106
175	1.0417323	359.067
176	1.0417382	360.028
177	1.0417441	360.989
178	1.0417500	361.950
179	1.0417559	362.911
180	1.0417618	363.872
181	1.0417677	364.833
182	1.0417736	365.794
183	1.0417795	366.755
184	1.0417854	367.716
185	1.0417913	368.677
186	1.0417972	369.638
187	1.0418031	370.599
188	1.0418090	371.560
189	1.0418149	372.521
190	1.0418208	373.482
191	1.0418267	374.443
192	1.0418326	375.404
193	1.0418385	376.365
194	1.0418444	377.326
195	1.0418503	378.287
196	1.0418562	379.248
197	1.0418621	380.209
198	1.0418680	381.170
199	1.0418739	382.131
200	1.0418798	383.092
201	1.0418857	384.053
202	1.0418916	385.014
203	1.0418975	385.975
204	1.0419034	386.936
205	1.0419093	387.897
206	1.0419152	388.858
207	1.0419211	389.819
208	1.0419270	390.780
209	1.0419329	391.741
210	1.0419388	392.702
211	1.0419447	393.663
212	1.0419506	394.624
213	1.0419565	395.585
214	1.0419624	396.546
215	1.0419683	397.507
216	1.0419742	398.468
217	1.0419801	399.429
218	1.0419860	400.390
219	1.0419919	401.351
220	1.0419978	402.312
221	1.0420037	403.273
222	1.0420096	404.234

品名: 普通水泥 规格: 42.5 品牌: 海螺 数量: 1000 吨

14. 2000-2001  
100% INCOME TAX - TDSILE

# 201-190-19 TITANIUM

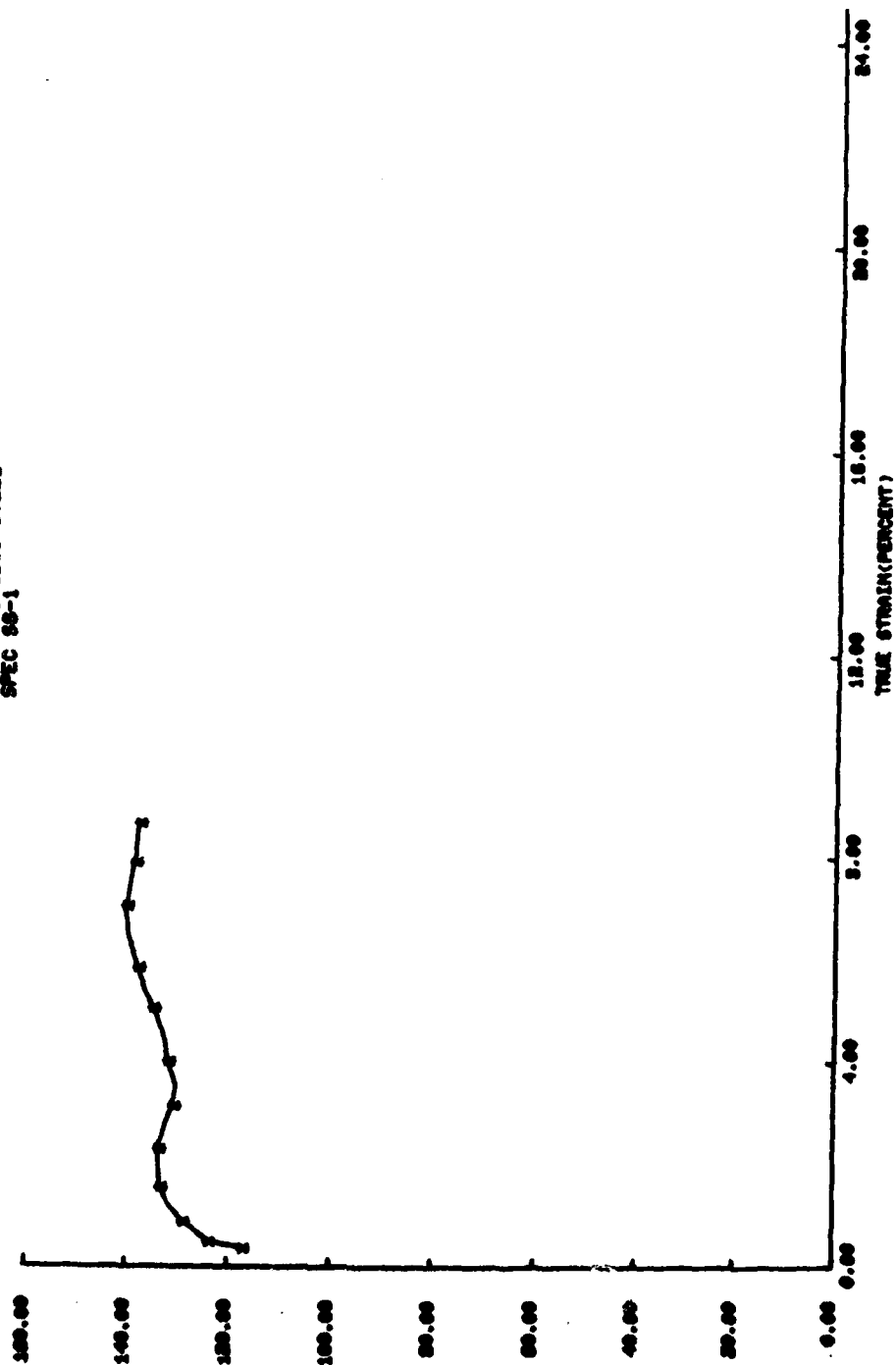
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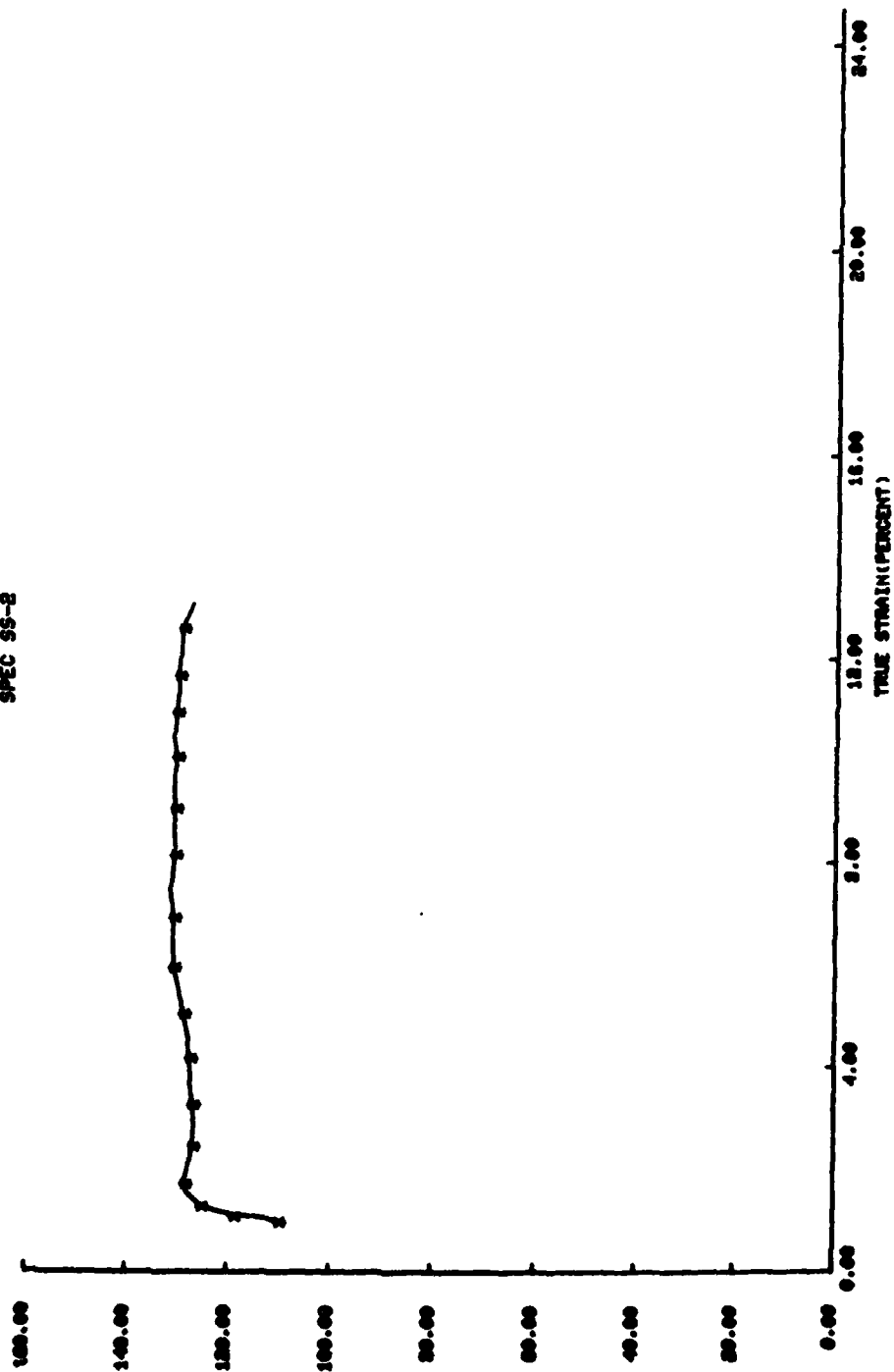
**● 附 錄 ●**

HOPKINSON BAR--TENSILE  
 340 STRAIN/SEC  
 410 STAINLESS STEEL  
 SPEC 90-1



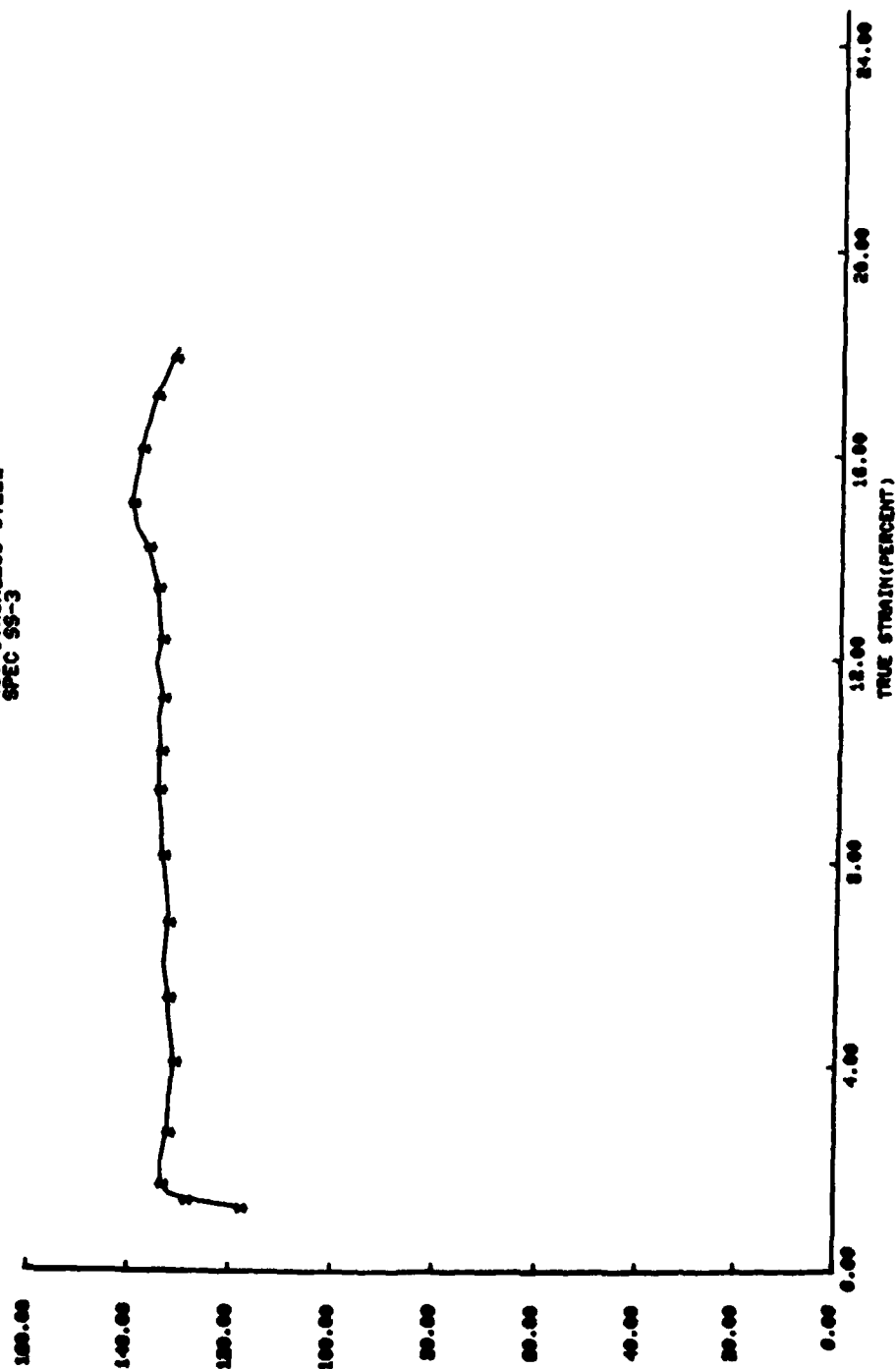
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HOPKINSON BAR--TENSILE  
 530 STRAIN/SEC  
 410 STAINLESS STEEL  
 SPEC SS-B



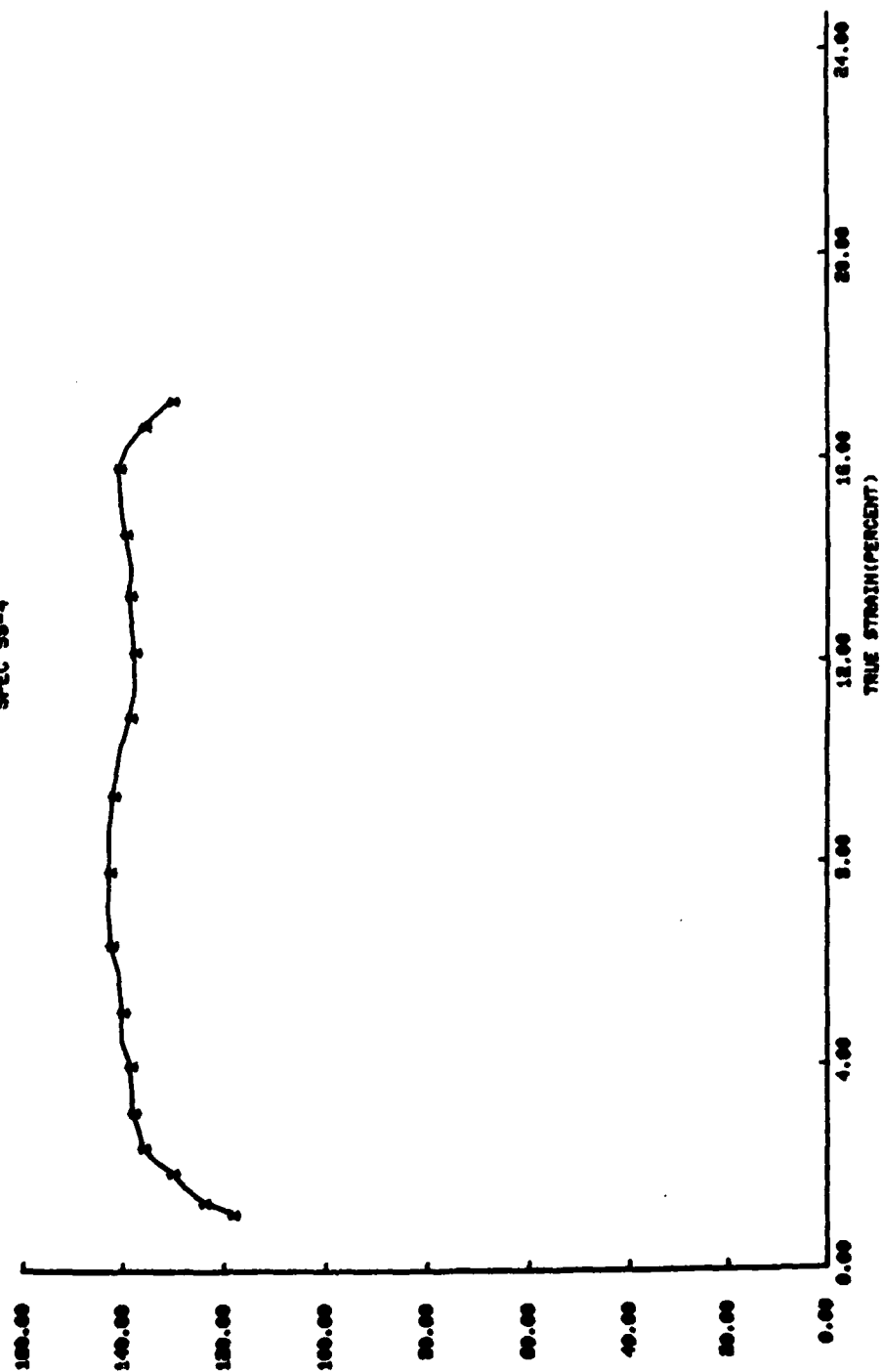
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HOPKINSON BAR--TENSILE  
 700 STRAIN/SEC  
 410 STAINLESS STEEL  
 SPEC 99-3



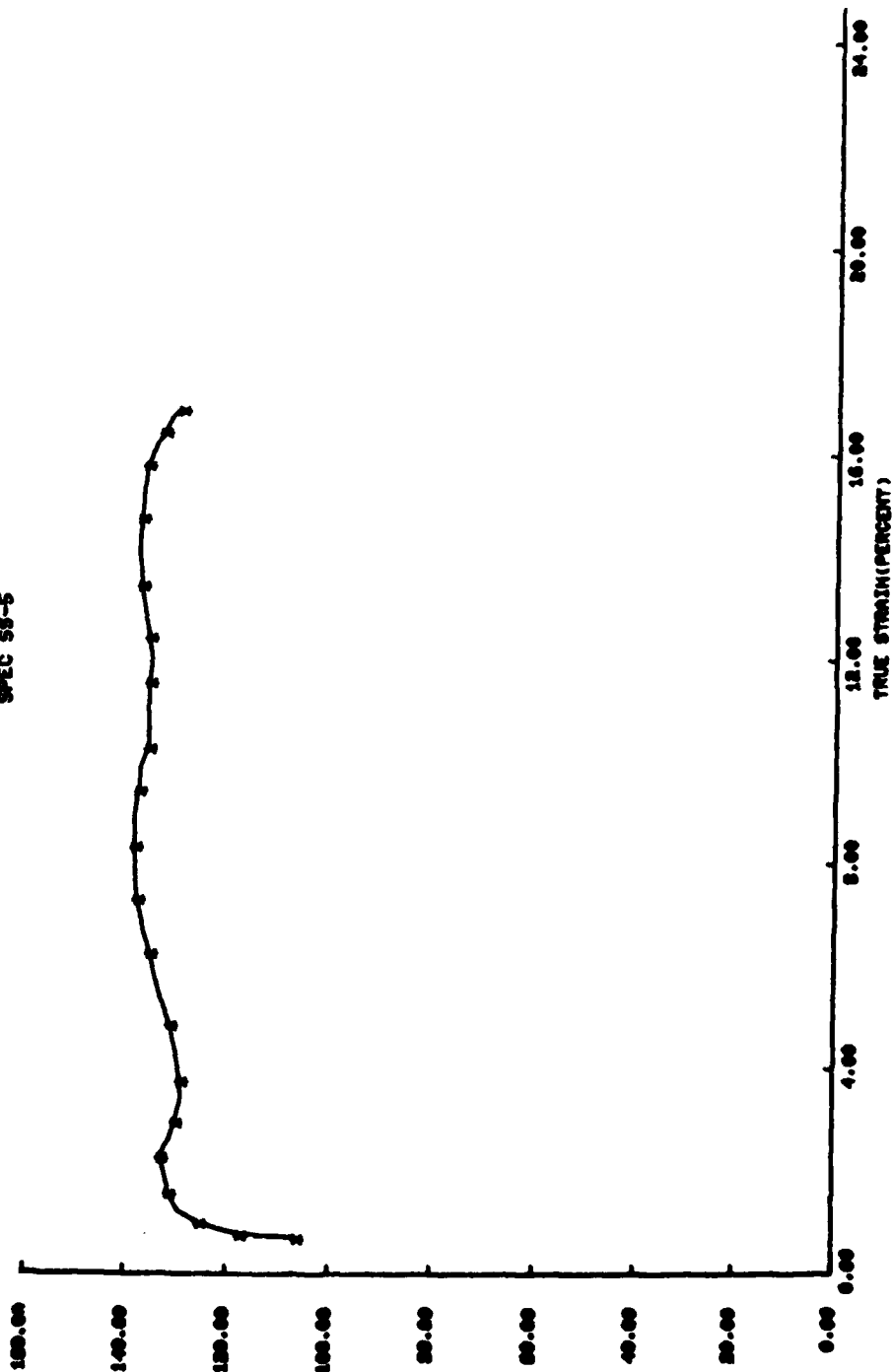


HOPKINSON BAR--TENSILE  
 769 STRAIN/SEC  
 416 STAINLESS STEEL  
 SPEC 58-4



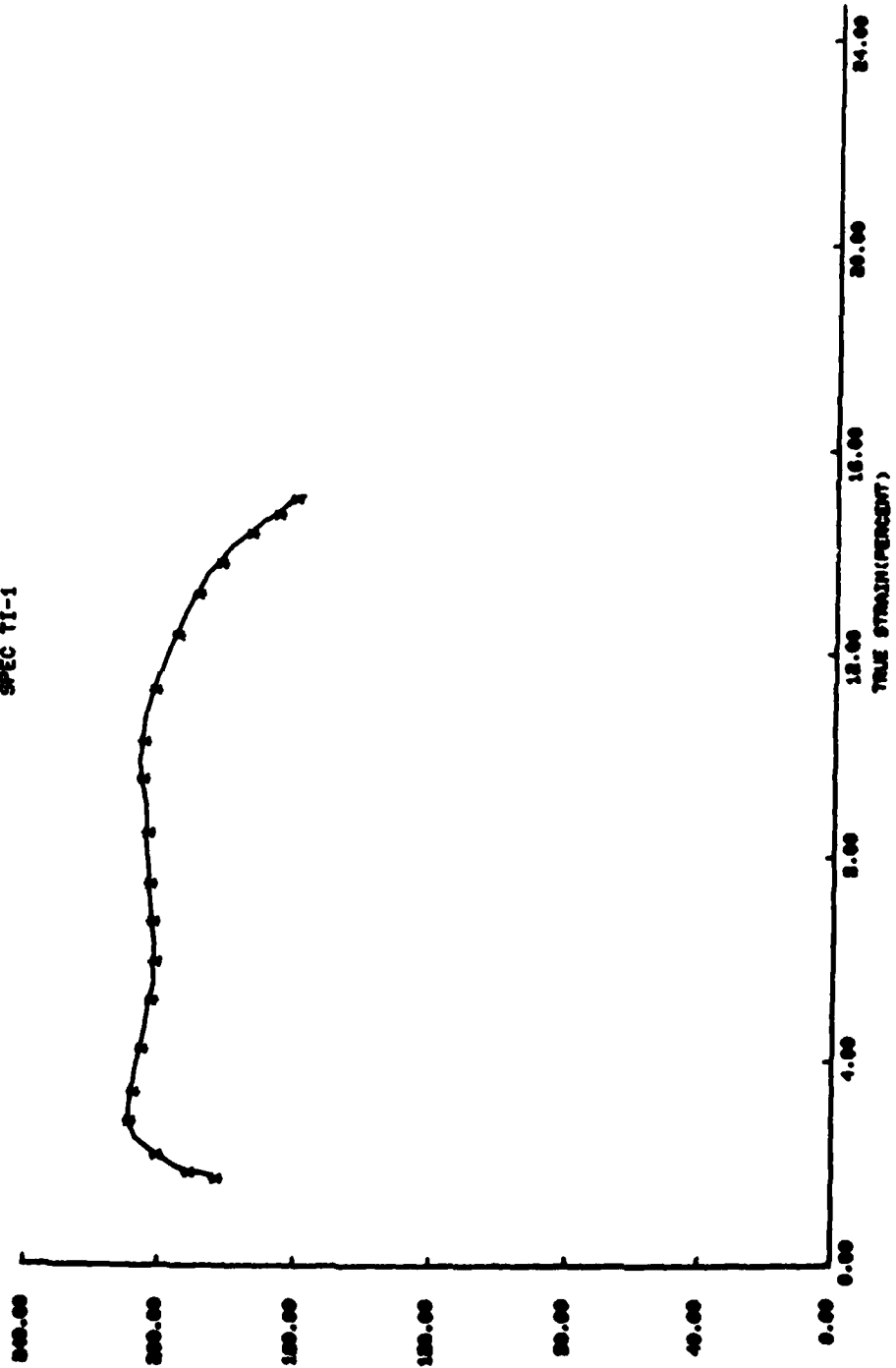
145 KSI

HOPKINSON BAR--TENSILE  
 600 STRAIN/SEC  
 410 STAINLESS STEEL  
 SPEC S8-5

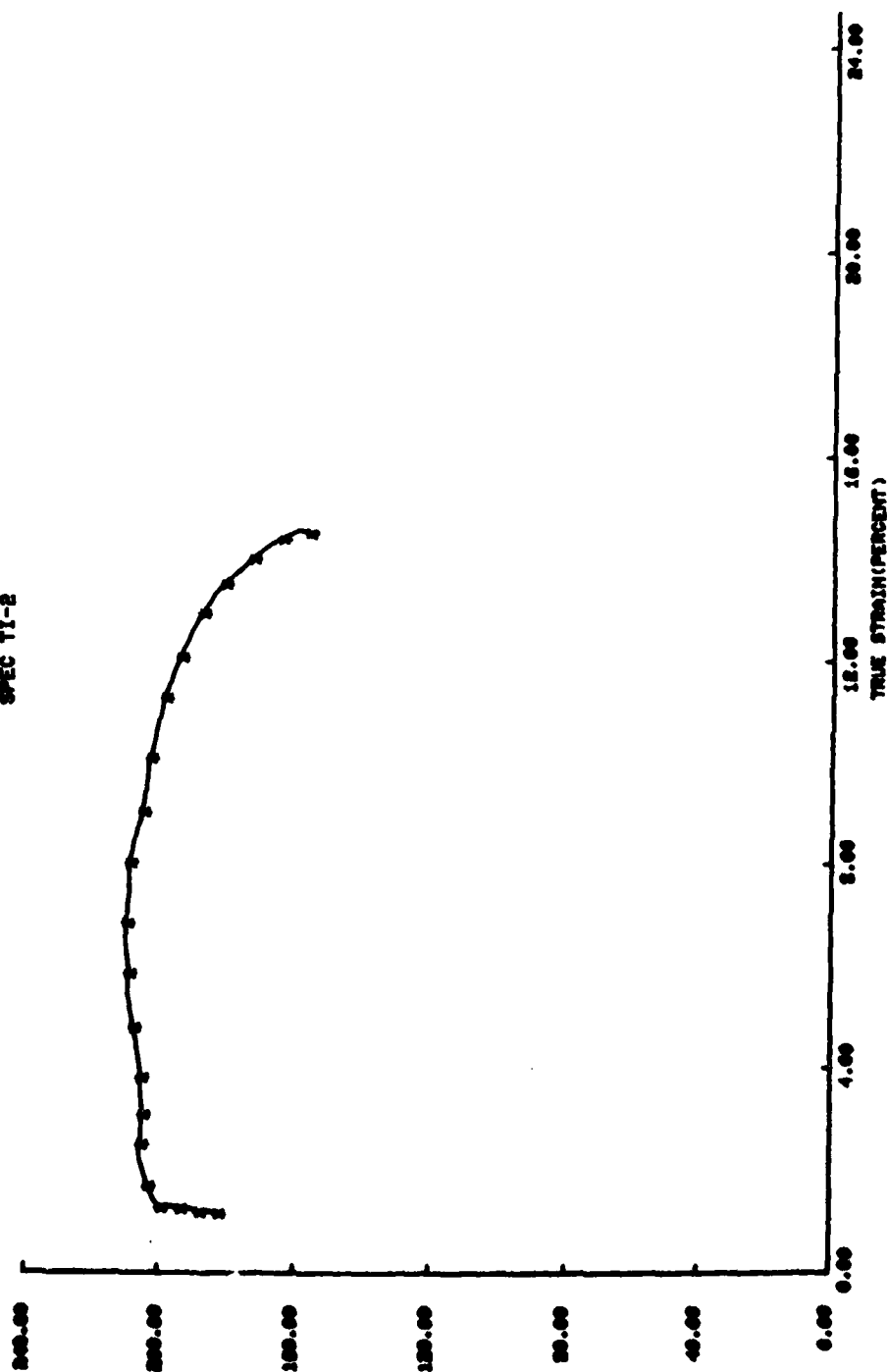


1-2-20-20 0-2-20-20-20 1-2-20-20

HOPKINSON BAR--TENSILE  
 500 STRAIN/SEC  
 BAL-INO-10 TITANIUM  
 SPEC TT-1

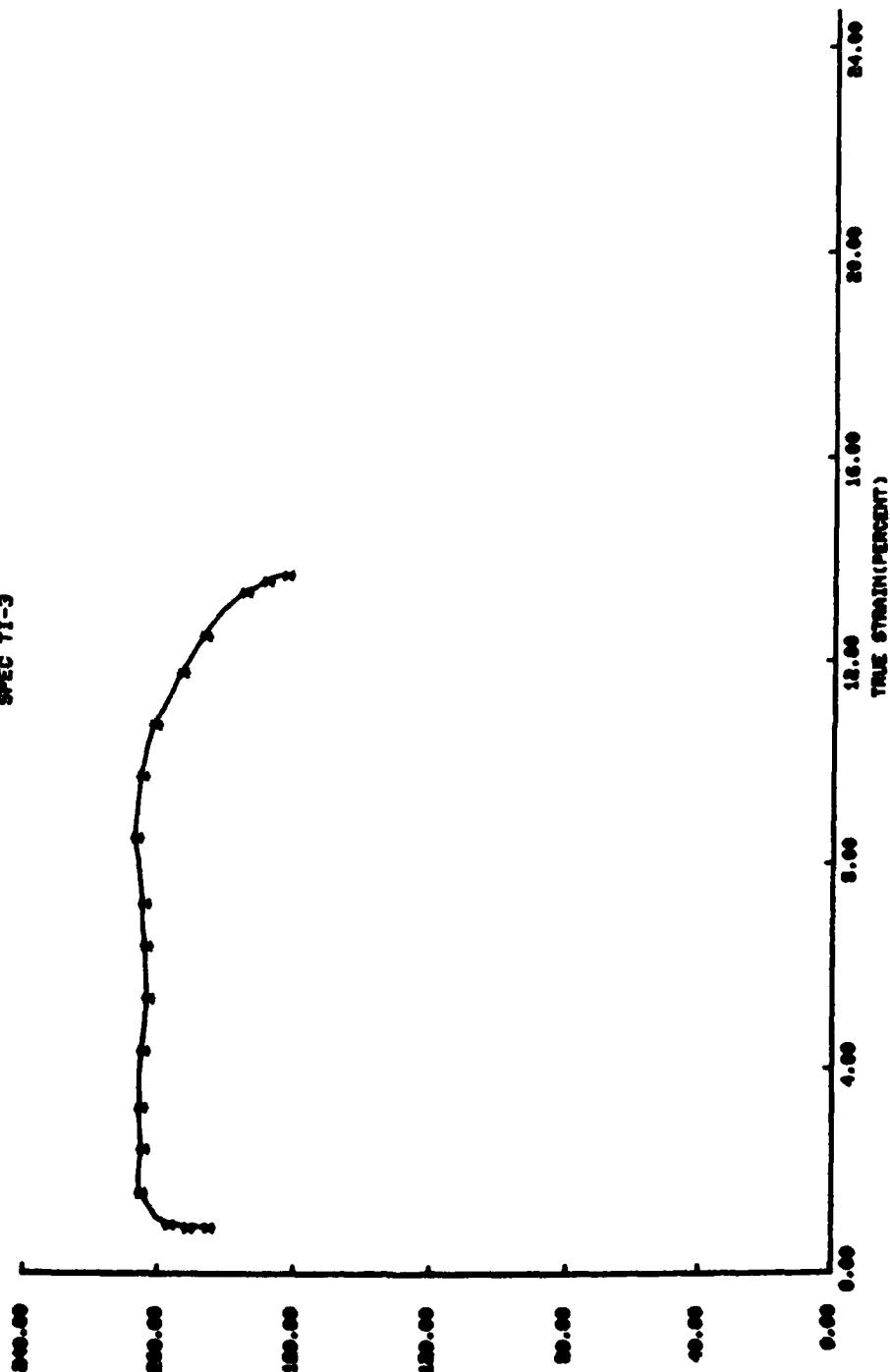


HOPKINSON BAR--TENSILE  
 500 STRAIN/SEC  
 BAL-100-10 TITANIUM  
 SPEC TI-2



7-22-64 00-000000-1 100-10

HOPKINSON BAR--TENSILE  
 550 STRAIN/SEC  
 BAL-100-10 TITANIUM  
 SPEC TI-3



1-10-68 10:00 AM

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82